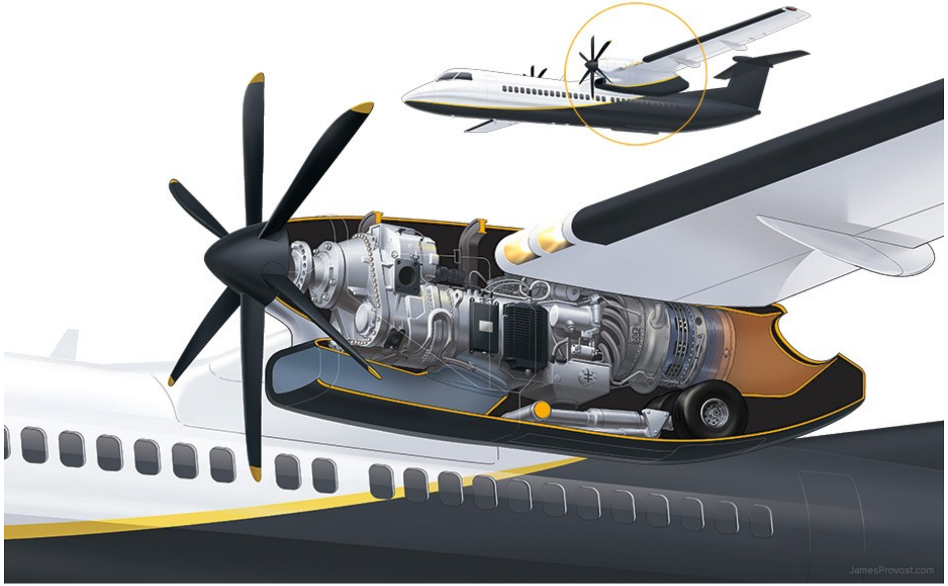
# Introduction to airbreathing propulsion systems

***APRI0004:***

***Integrated project aerospace design***



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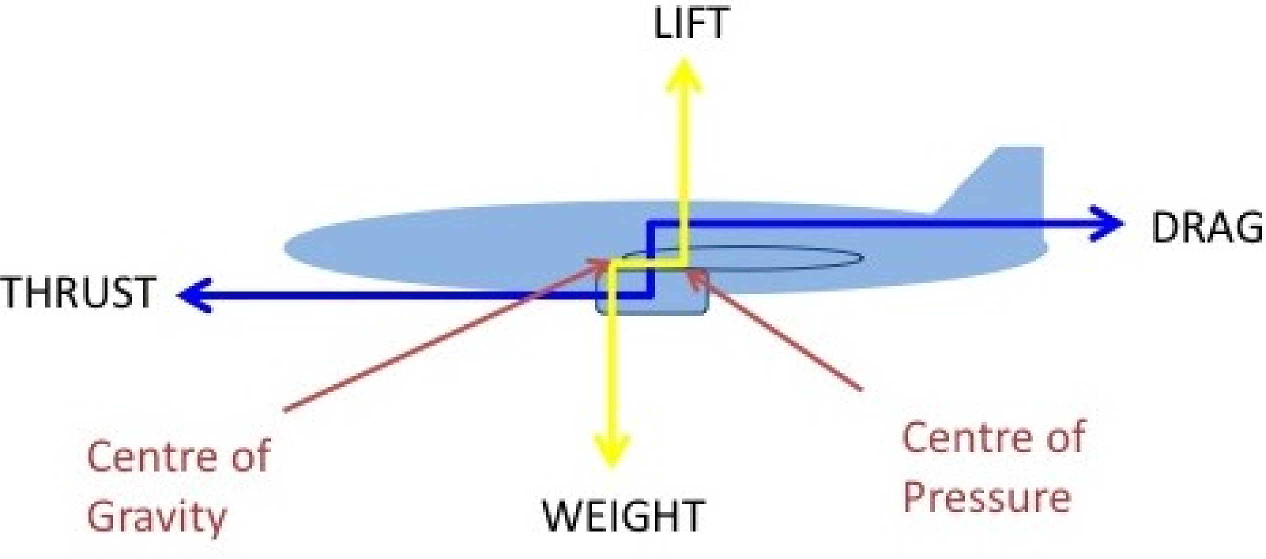
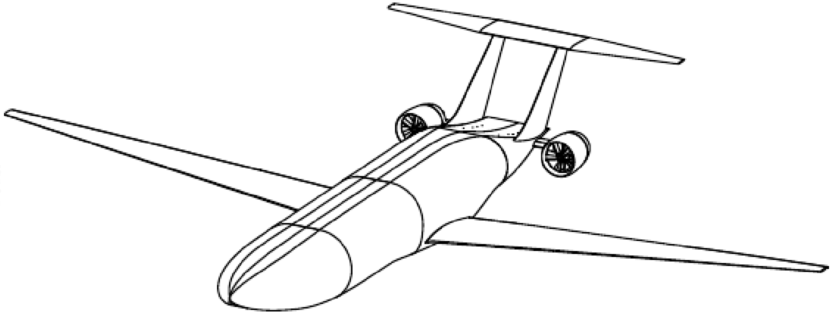
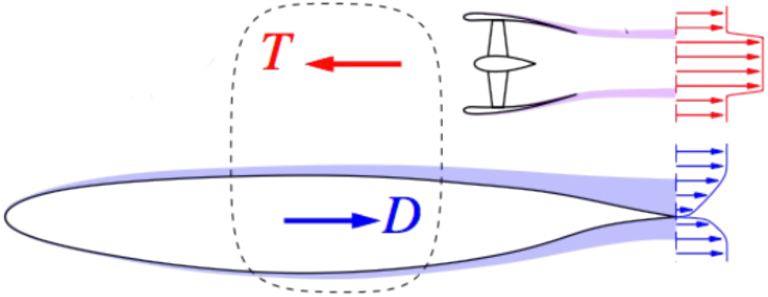
# 1. Balances, thrust and performance

*Drag / thrust definition*





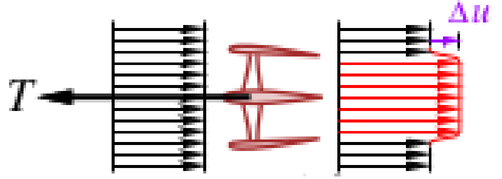
# 1. Balances, thrust and performance

*Airbreathing engines: acceleration of (clean) air mass flow*

- **Thrust = acceleration force of engine mass flow from flight $v_f$ to jet velocity $v_j$**

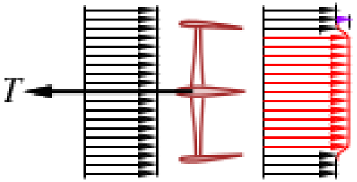
$$\mathcal{T} = \dot{m}_a (v_j - v_f)$$

- **Powers**
  - Propulsive power → airplane acceleration : $\mathcal{P}_p = \mathcal{T} u_f = \dot{m}_a (u_j - u_f) u_f = \dot{m}_a \, \Delta u \, u_f$
  - Mechanical power → fluid acceleration : $\mathcal{P}_m = \dot{m}_a \Delta \mathcal{E}_k = \dot{m}_a \dfrac{u_j^2 - u_f^2}{2}$
  - Lost power $\mathcal{P}_l = \mathcal{P}_m - \mathcal{P}_p = \dot{m}_a \dfrac{\Delta u^2}{2}$

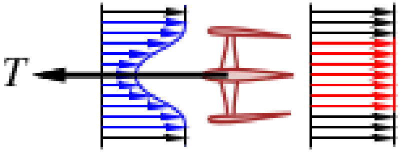


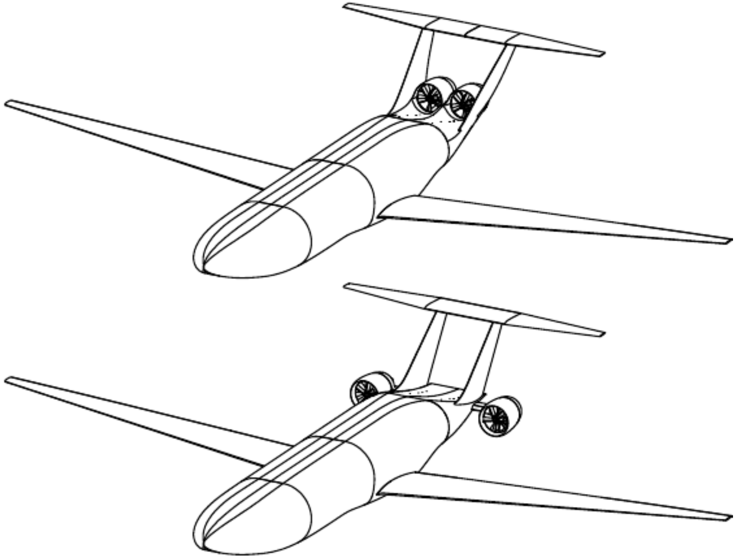

- **Propulsive efficiency:**

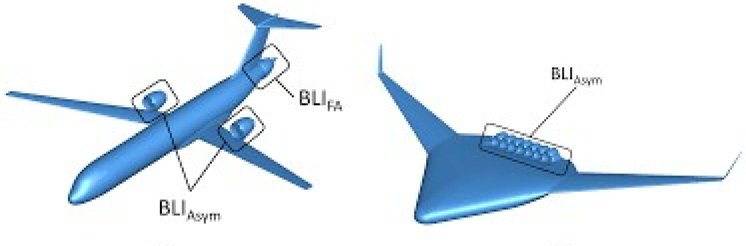
$$\eta_p = \frac{\mathcal{P}_p}{\mathcal{P}_m} = \frac{2 u_f}{u_f + u_j}$$



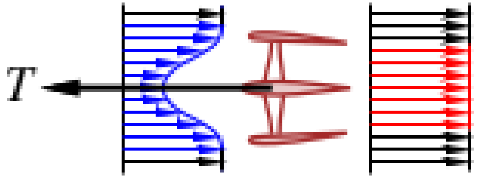

- **Increasing propulsive efficiency for constant thrust:**
  - increase mass flow, decrease jet velocity
  - Ingest flow at speed lower than flight speed $\mathcal{T} = \dot{m}_a (v_f - v_i)$



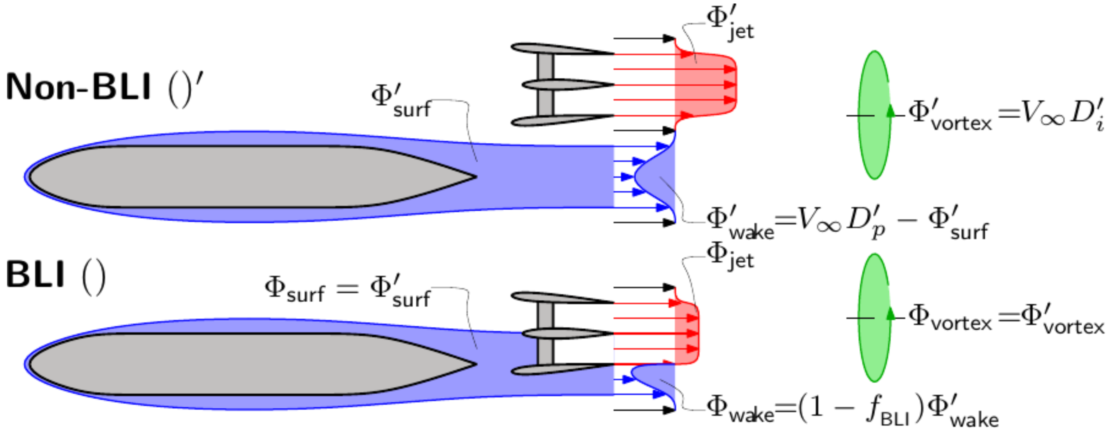





(a) (b)




$$\mathcal{T} = \dot{m}_a(v_j - v_i)$$

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# 1. Balances, thrust and performance

*Airbreathing engines: classical performance parameters*

- **Thermal energy $Q = m_f \Delta h_f$**
  - Fuel mass flow rate: $\dot{m}_f$
  - Fuel to air ratio: $far = \dot{m}_f / \dot{m}_a$
  - Fuel lower heating value: $\Delta h_f \approx 43 MJ/kg$
- **Overall efficiency : propulsive power $P_p$ versus thermal energy Q**
  - Propulsive efficiency:

$$\eta_p = \frac{\mathcal{P}_p}{\mathcal{P}_m} = \frac{2v_f}{v_f + v_j}$$

  - Thermal efficiency:

$$\eta_t = \frac{\mathcal{P}_m}{\mathcal{Q}} = \frac{\mathcal{P}_m}{\dot{m}_f \Delta h_f}$$

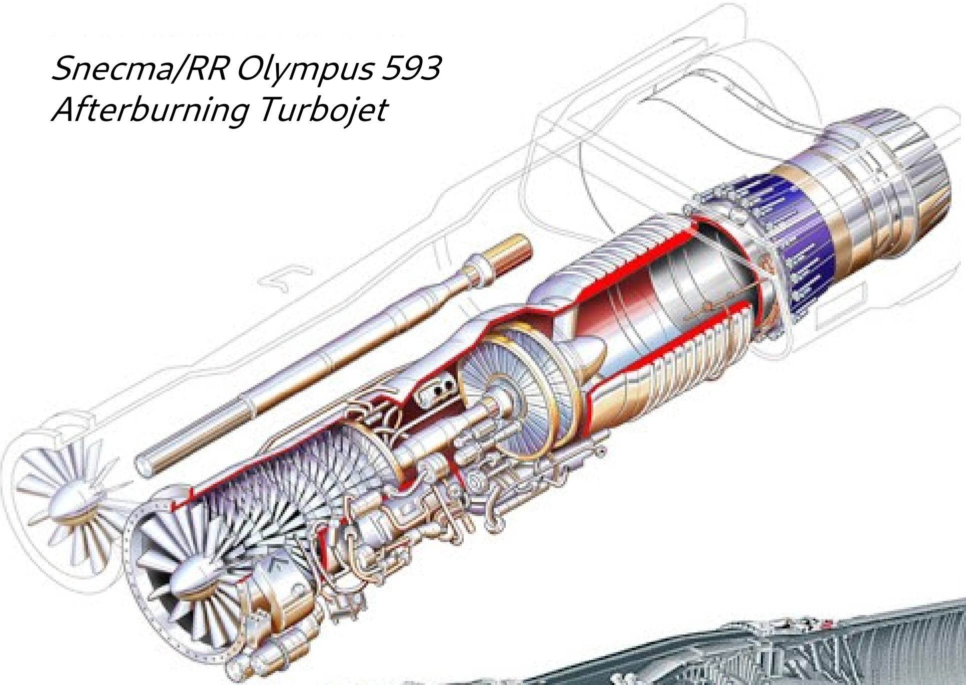
$$\eta = \frac{\mathcal{P}_p}{\mathcal{Q}} = \eta_p \eta_t$$

- **Efficiency → *Thrust specific fuel consumption (TSFC)*** $TSFC = \frac{\dot{m}_f}{\mathcal{T}}$
- **Compacity → *Specific thrust*** $\mathcal{T}_s = \frac{\mathcal{T}}{\dot{m}_a}$

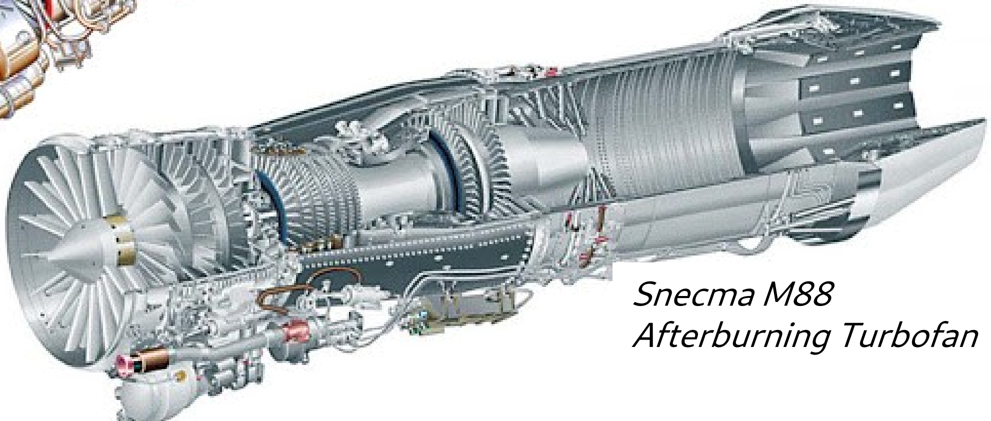
# 2. Jet engines

*Generation of “high” speed jet through expansion over nozzle*

*Snecma/RR Olympus 593*
*Afterburning Turbojet*



*CFM Leap*
*Civil Turbofan*



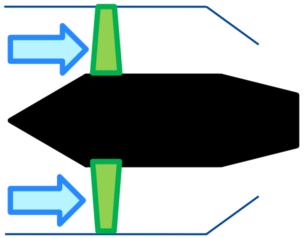
*Snecma M88*
*Afterburning Turbofan*

# 2. Jet engines

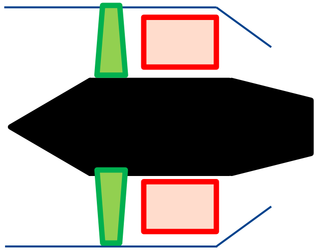
*Generation of "high" speed jet through expansion over nozzle*

- **Ingestion of $m_a$ air at flight speed in nacelle**
  - Ram effect: increased total T and p due to relative Mach number $M_f$

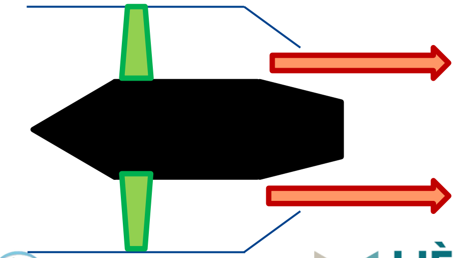
$$T^\circ = T_a \left(1 + \frac{\gamma - 1}{2} M_f^2\right) \qquad p^\circ = p_a \left(\frac{T^\circ}{T_a}\right)^{\frac{\gamma}{\gamma-1}}$$



- **Increase total pressure and temperature**
  - Mechanical : fan
  - Thermal : gas generator / Brayton
  - Afterburning



- **Expansion over exhaust nozzle to ambient pressure**
  - Choked
  - Adapted



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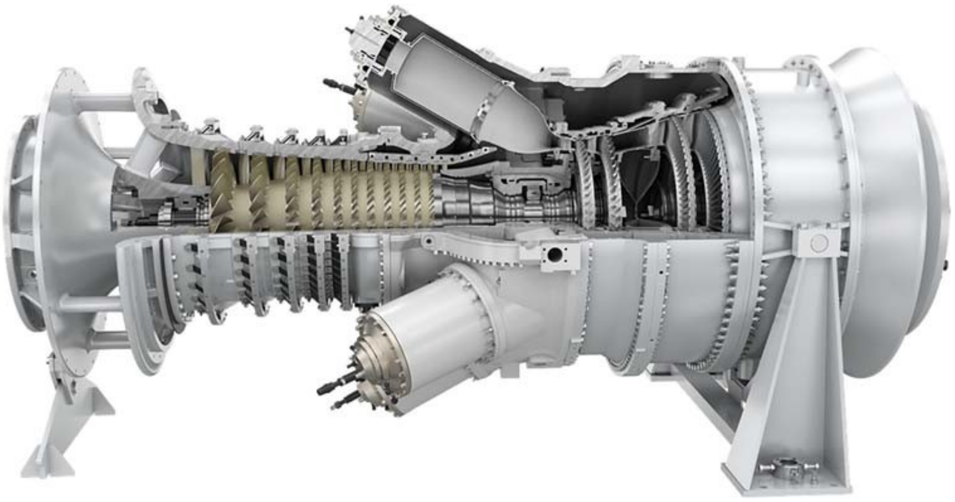
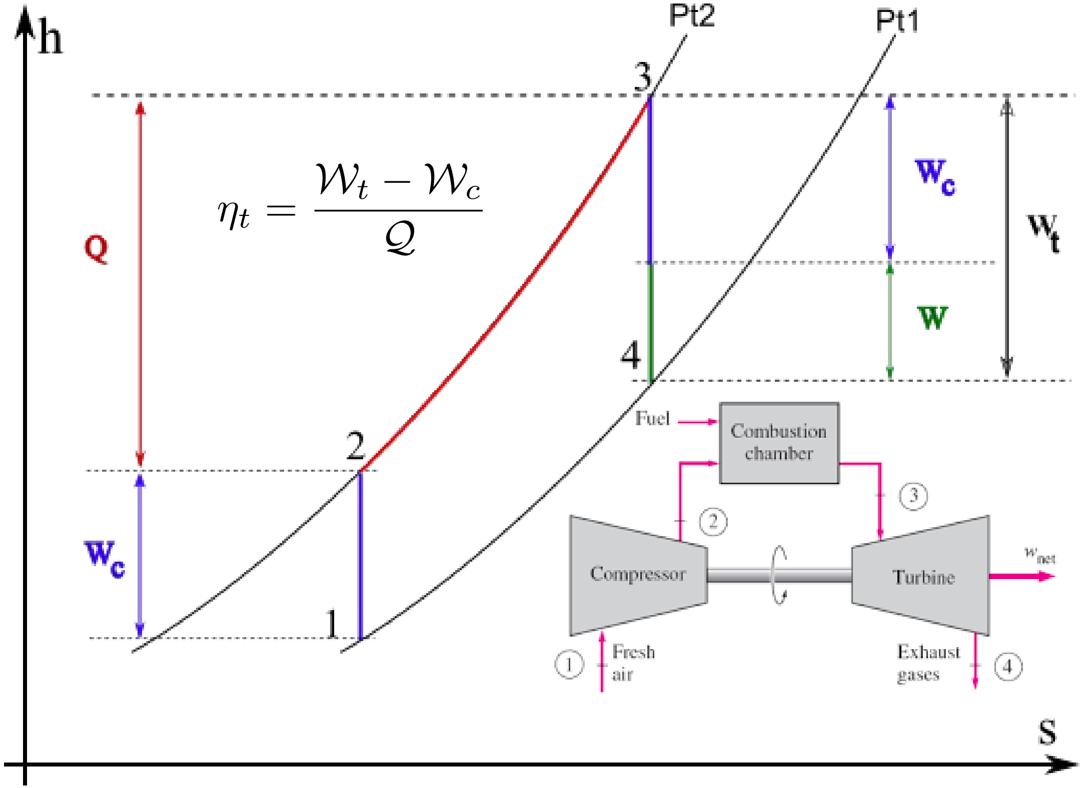
- **Thermodynamic cycle**
  - Adiabatic compression 1 → 2 : $W_c = \dot{m}\Delta H_{12}$
  - Combustion 2 → 3 : $Q = \dot{m}\Delta H_{23}$
  - Adiabatic expansion 3-4 : $W_t = \dot{m}\Delta H_{43}$





- **Parameters affecting efficiency**
  - Overal pressure ratio (OPR) : $\Pi = \dfrac{P_2^\circ}{P_1^\circ}$
  - Turbine Inlet Temperature (TIT)/ $T_3^\circ$

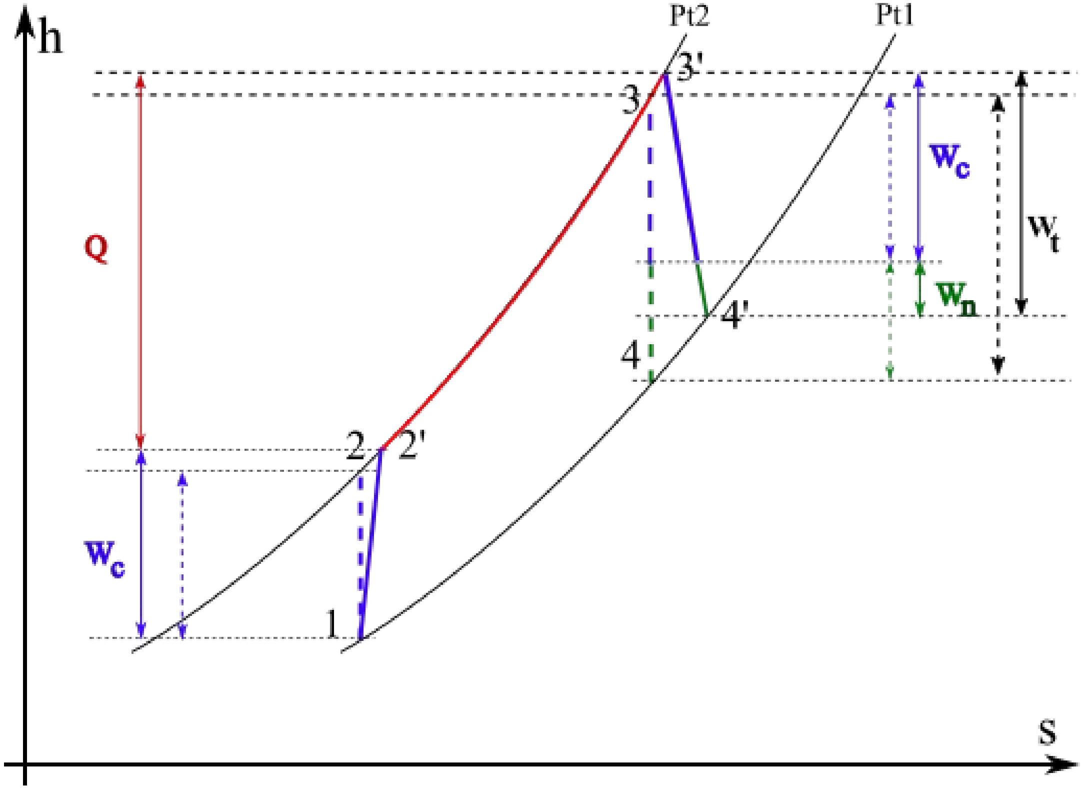
    Turbine Entry Temperature (TET)

    *or* Overal Temperature ratio: $\tau = \dfrac{T_3^\circ}{T_1^\circ}$
  - Compressor efficiency

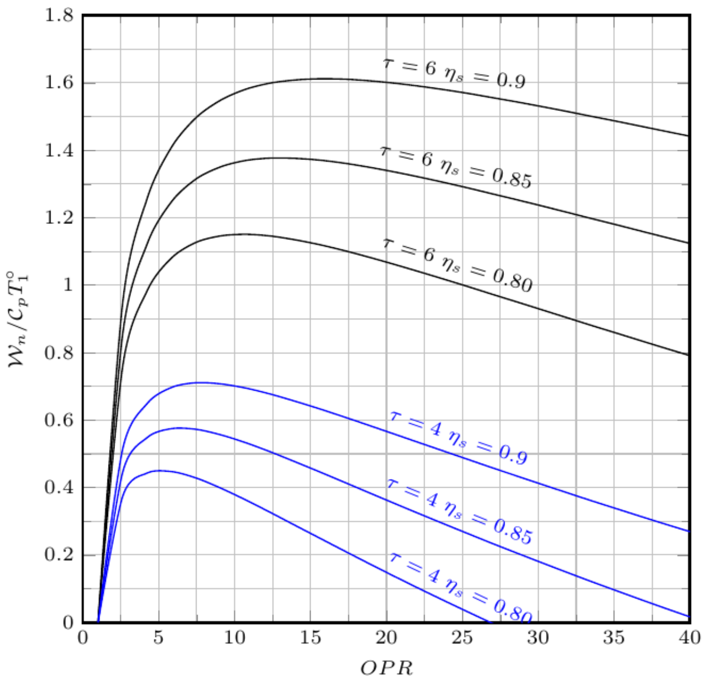
$$\eta_c = \frac{h_2^\circ - h_1^\circ}{h_{2'}^\circ - h_1^\circ}$$

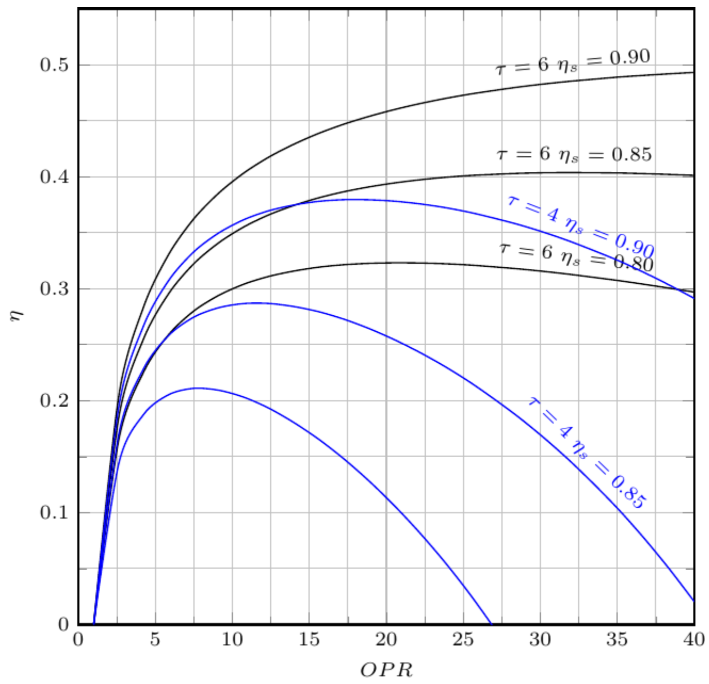
  - Turbine efficiency

$$\eta_t = \frac{h_3^\circ - h_{4'}^\circ}{h_3^\circ - h_4^\circ}$$



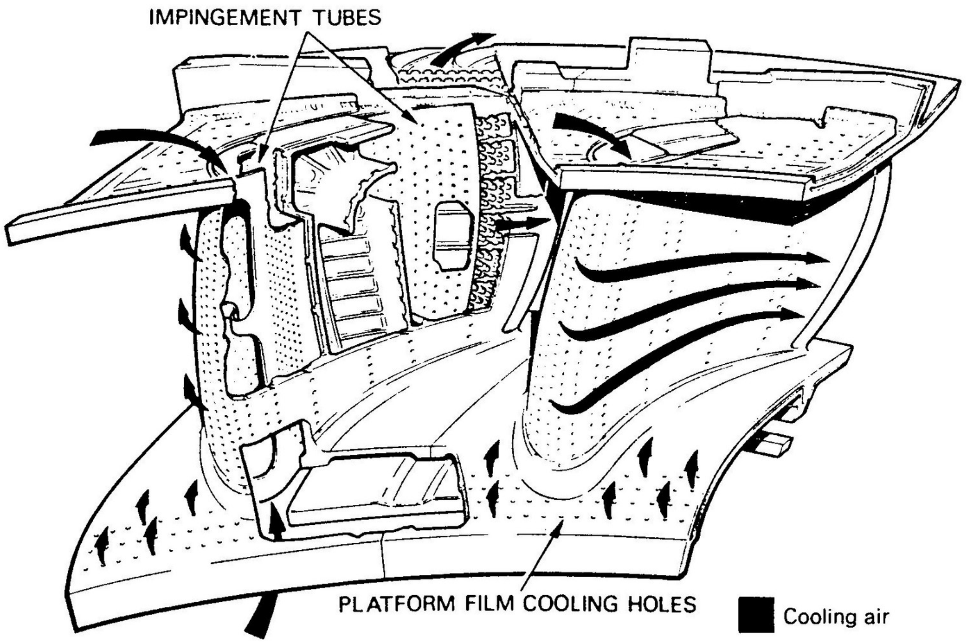
- **TiT**
  - Determines specific work
  - Limited by material resistance
  - ~ 1850 K >> 1400 K (fusion)
  - Requires film cooling
-
- **OPR**
  - Determines efficiency
  - Optimum ~ TiT & efficiencies
  - 30 … 40

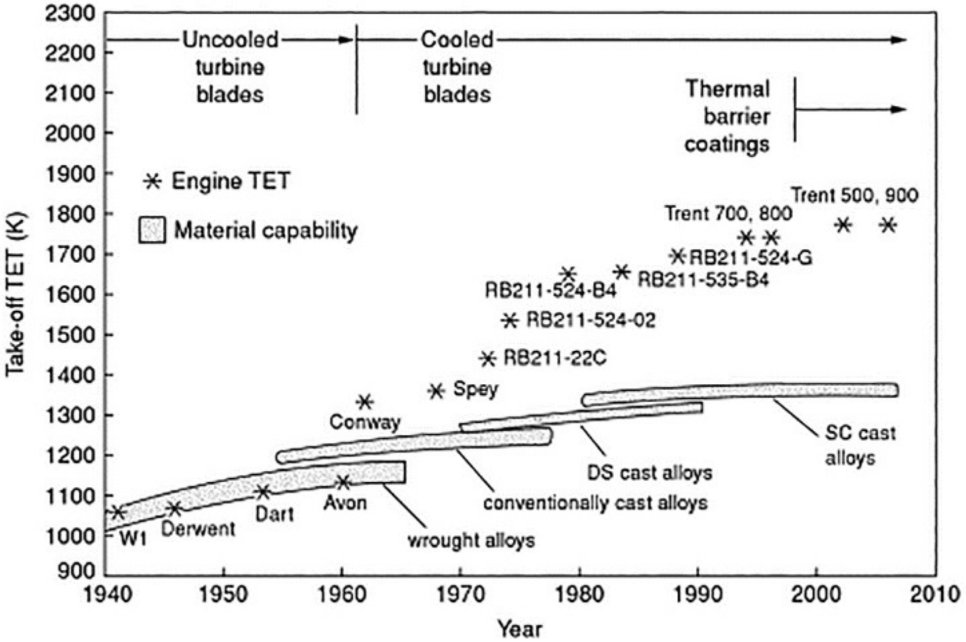


# 2. Jet engines

*Core flow : Increasing TiT / TeT*



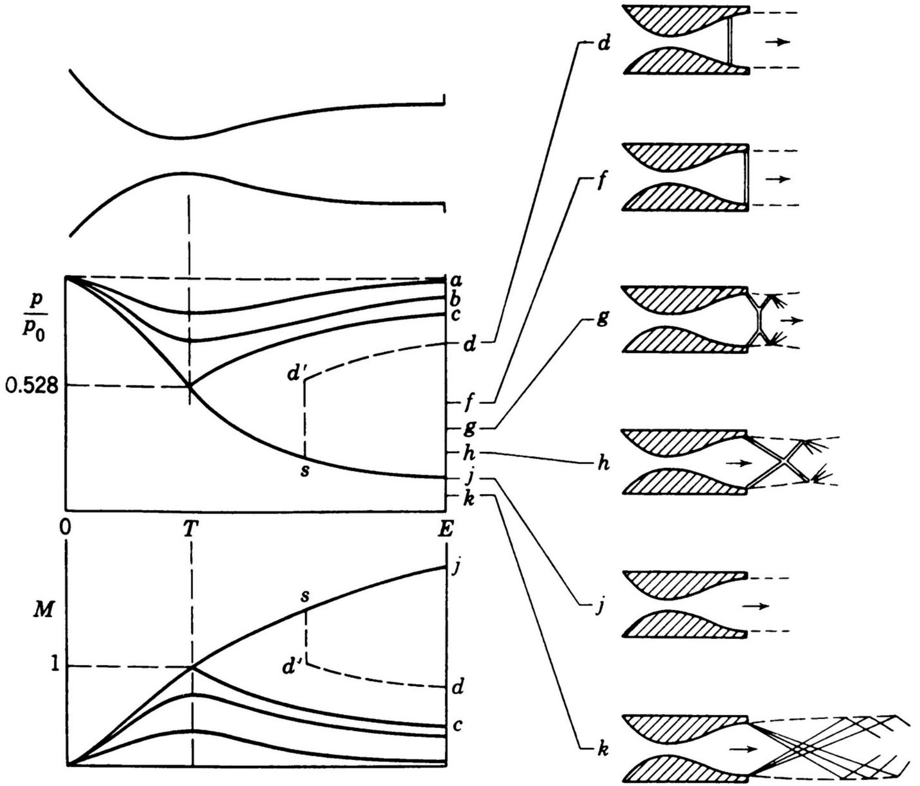



# 2. Jet engines
*Nozzle*

- **Jet engine nozzle ~ de Laval nozzle**
  - Nozzle pressure ratio $NPR = \dfrac{p^\circ}{p_a}$
  - Critical pressure ratio ~ choking $NPR^* = \dfrac{p^\circ}{p^*} \approx 1.78$
  - Choking mass flow rate $\dot{m}^* \approx 0.68 \dfrac{p^\circ A}{\sqrt{RT^\circ}}$
- **Thrust for imperfect expansion**

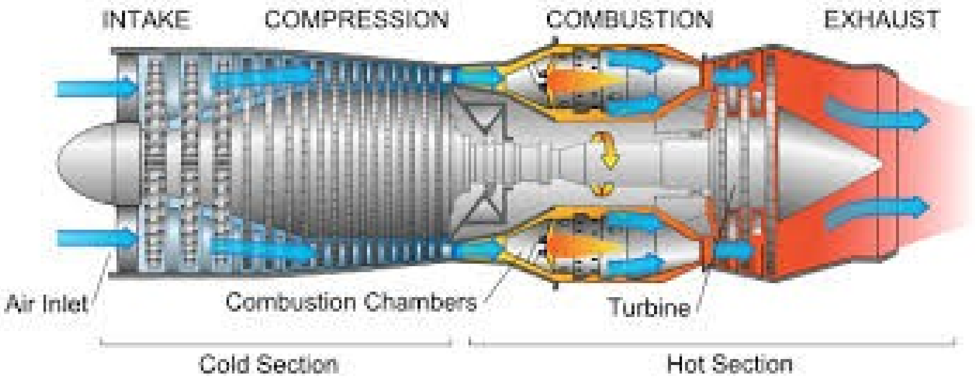
$$T = \dot{m}(v_j - v_f) + (p_j - p_a)A$$

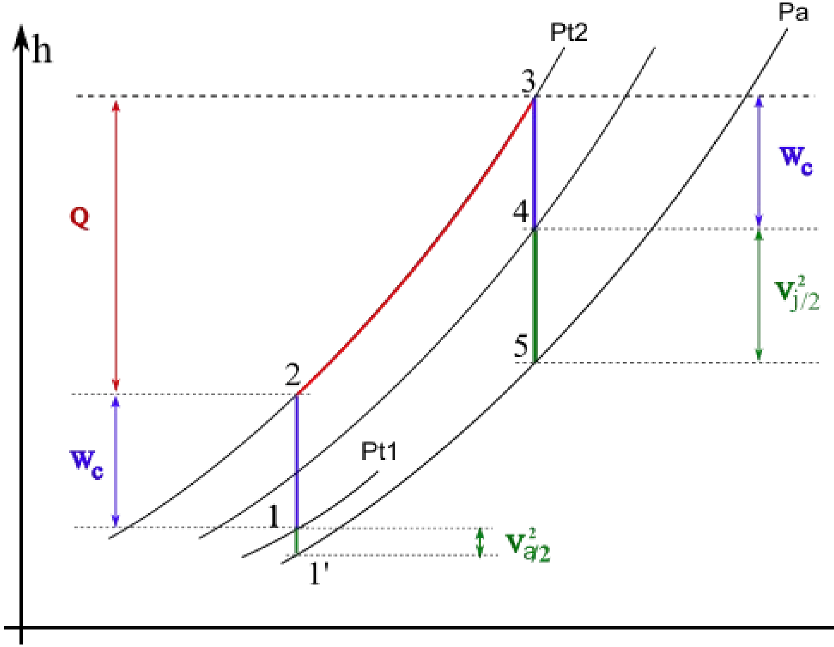
- **Maximal thrust if *adapted* ($p_j = p_a$)** $T = \dot{m}(v_j - v_f)$
- **Engine operating point = match GG and nozzle**
  - GG determines p° and T° upstream of nozzle
  - Nozzle limits maximum mass flow rate
  - Thrust can be optimized by varying throat (and expansion ratio)
  - Area needs to be variable to accommodate large variations in T°



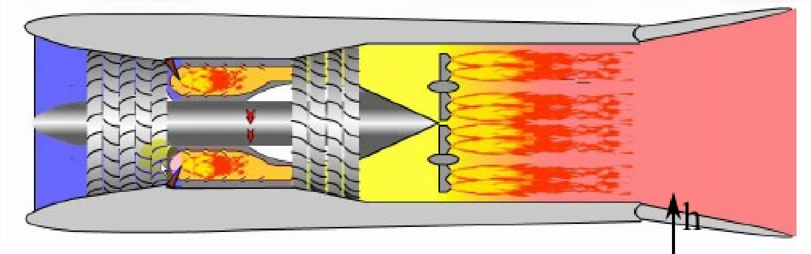
## *Turbojet : high subsonic through supersonic flow*

- **Core flow**
  - RAM effect : 1' - 1
  - Compressor : 1 – 2
  - Combustion : 2 – 3
  - Turbine expansion : 3 – 4
  - Exhaust nozzle jet : 4 – 5
- **High specific thrust**

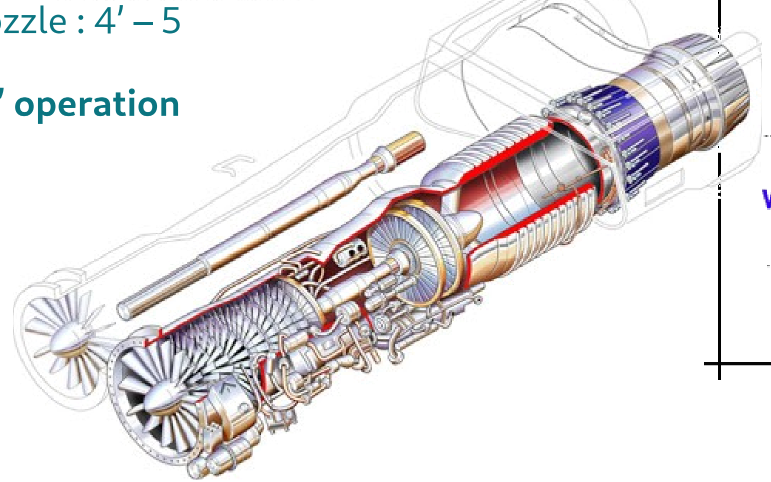


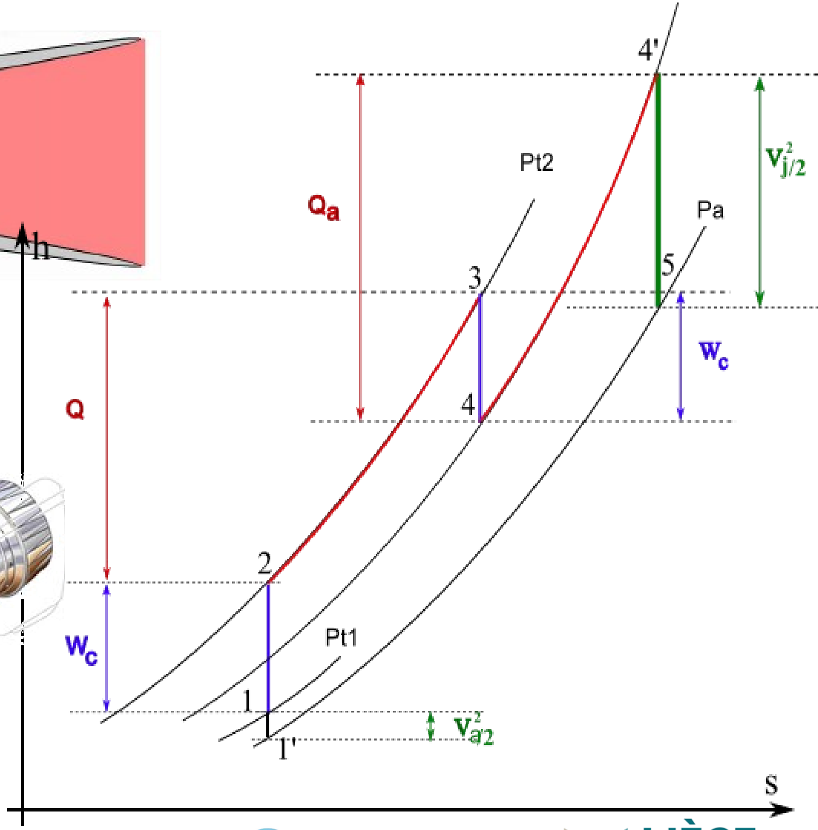



- **Core flow**
  - Ram effect : 1′ - 1
  - Compressor : 1 – 2
  - Combustion : 2 – 3
  - Expansion in turbine  : 3 – 4
  - **Wet / afterburning** : 4 – 4′
  - Exhaust by adjustable nozzle : 4′ – 5
- **Nearly double thrust in “wet” operation**



ROLLS ROYCE OLYMPUS 593 MRK 610 TURBOJET



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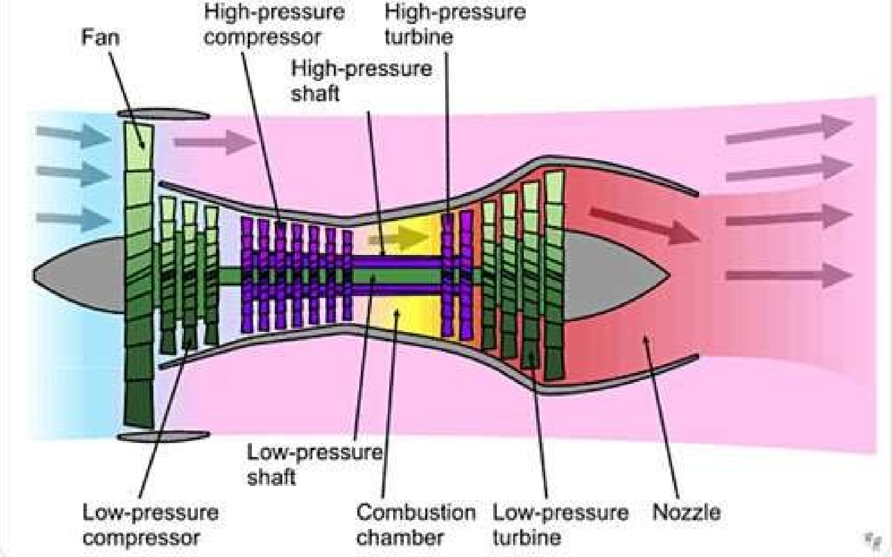
# 2. Jet engines
## *Civil turbofan : high subsonic/transonic*

- **Core $m_c$ and bypass $m_b$ flow rate**
  - Bypass ratio (BPR): $\alpha = m_b/(m_b+m_c)$
- **Core / primary : mech. power**
  - Fan + compressor : 1 - 2
  - Combustion : 2 - 3
  - Turbine → fan & compressor 3 - 4
  - Exhaust jet : 4 – 5




- **Bypass / secondary flow : thrust**
  - Compression by fan 1 – 2'
  - Exhaust nozzle 2' - 5

- **High propulsive efficiency: small acceleration of high mass flow rate**

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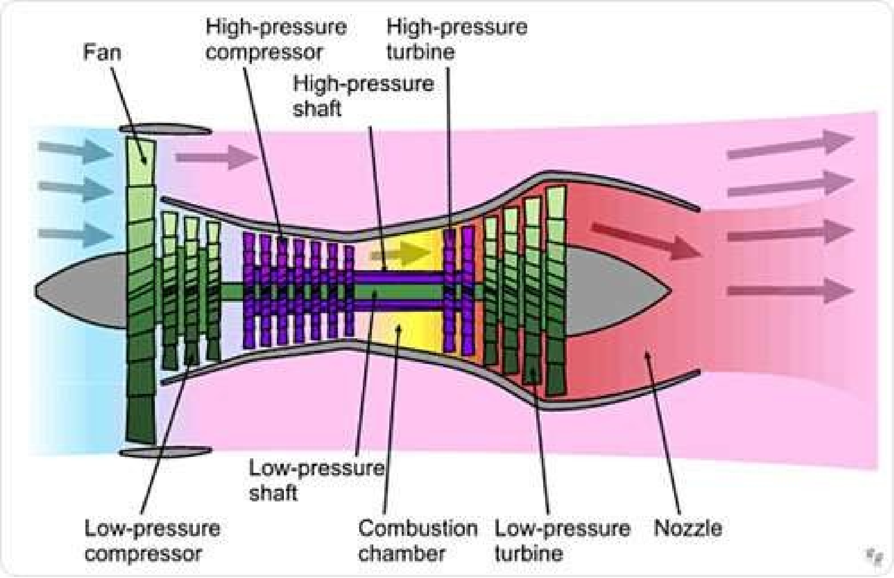
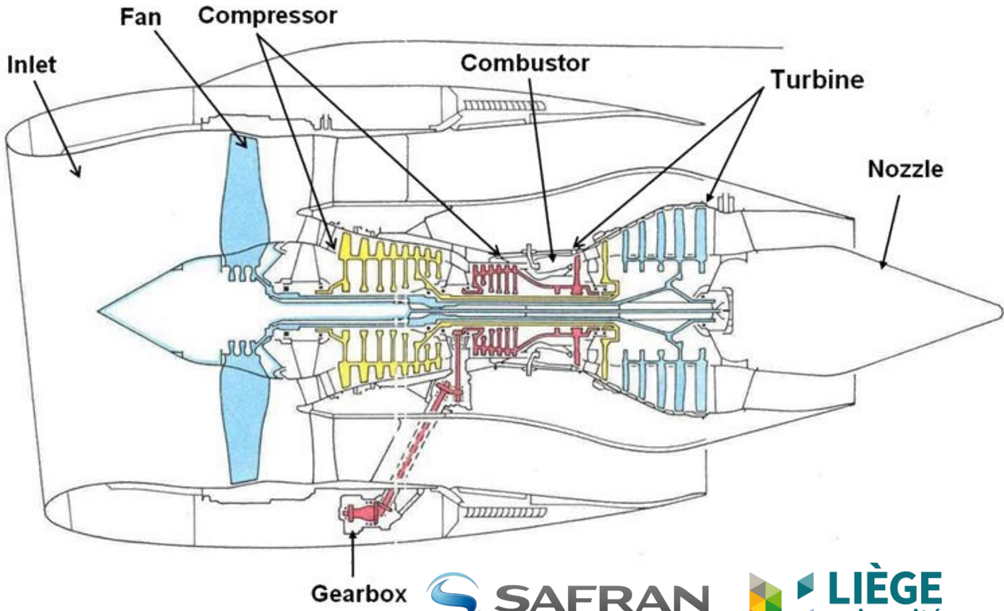
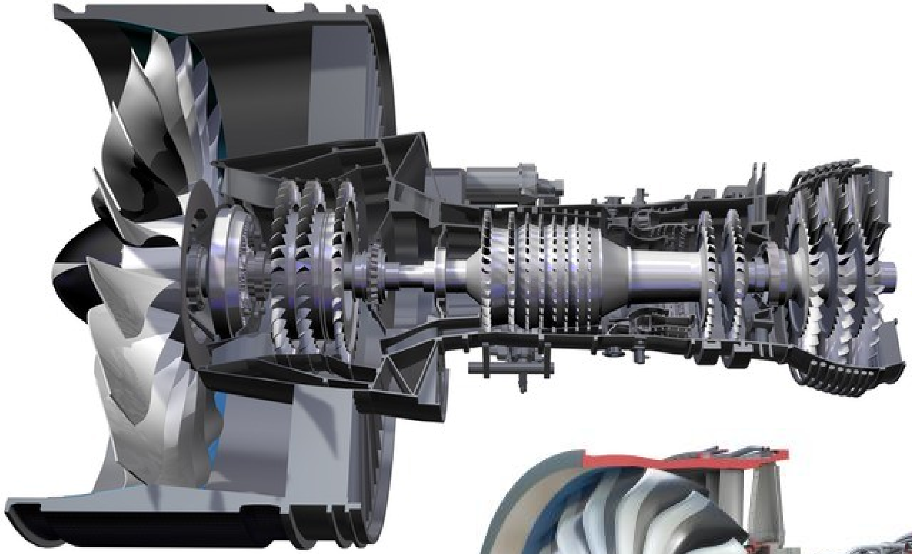
# 2. Jet engines
## *Civil turbofan: multispool turbofans*

- Optimal rotation speed = slightly supersonic at the tip
- Different spools / shafts → optimise rotation speed
- LP spool drives fan → large LP turbine



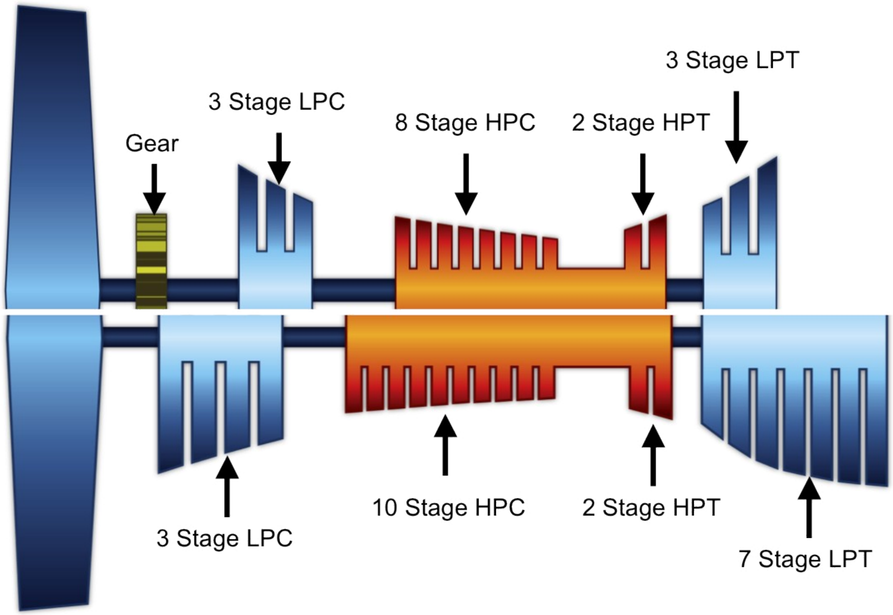
# 2. Jet engines
*Civil turbofan : geared turbofan (GTF)*



PW1100G
BPR = 12.5

CFM Leap
BPR = 9 .. 11

PW1100G

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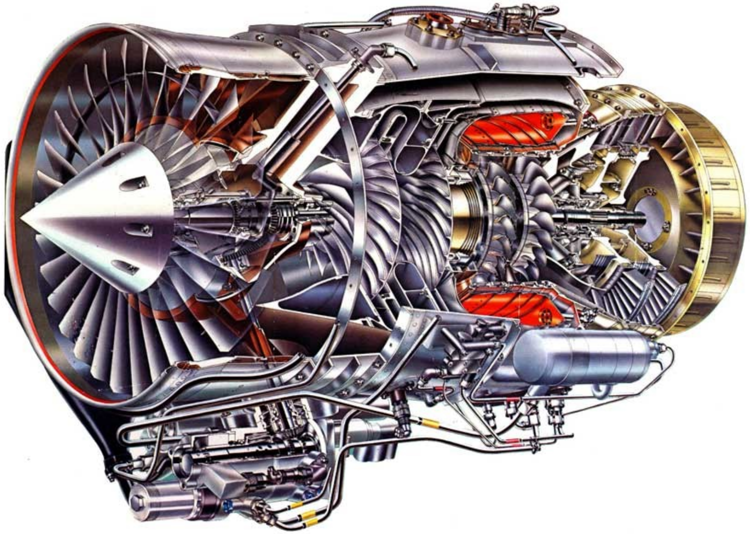
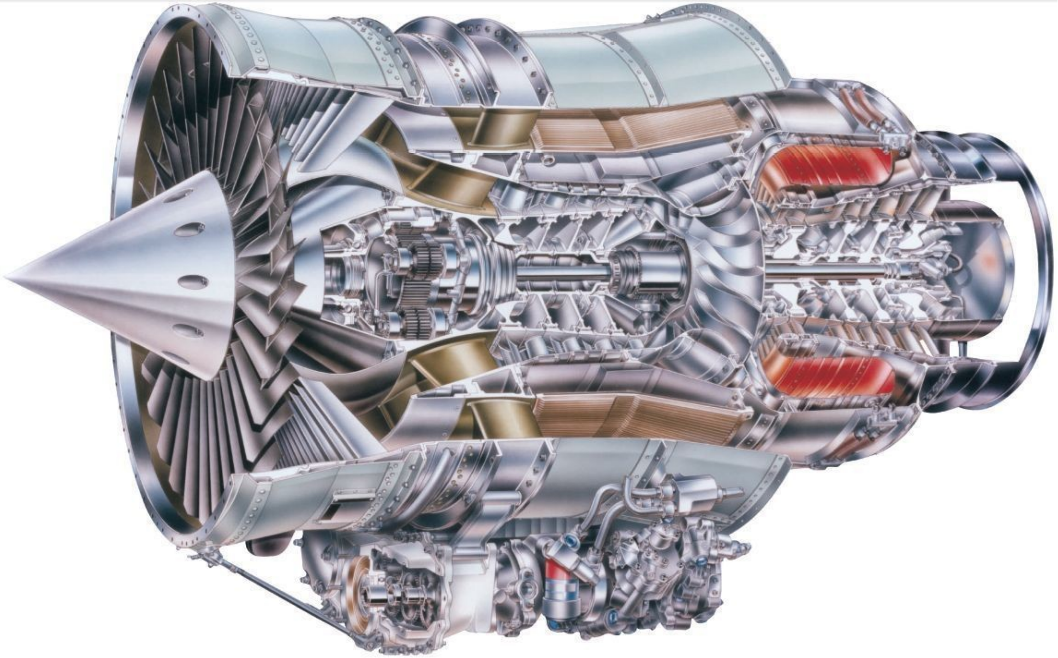
# Bizjet turbofans



Garret F109
T ~ 6kN
BPR ~ 5

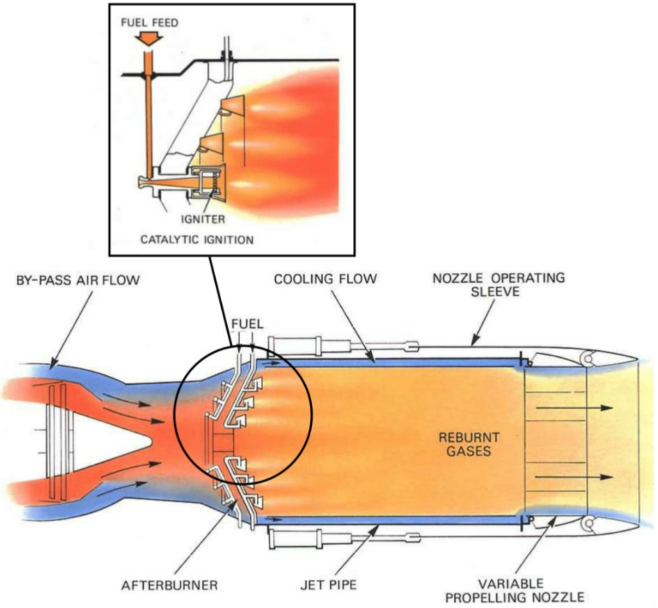


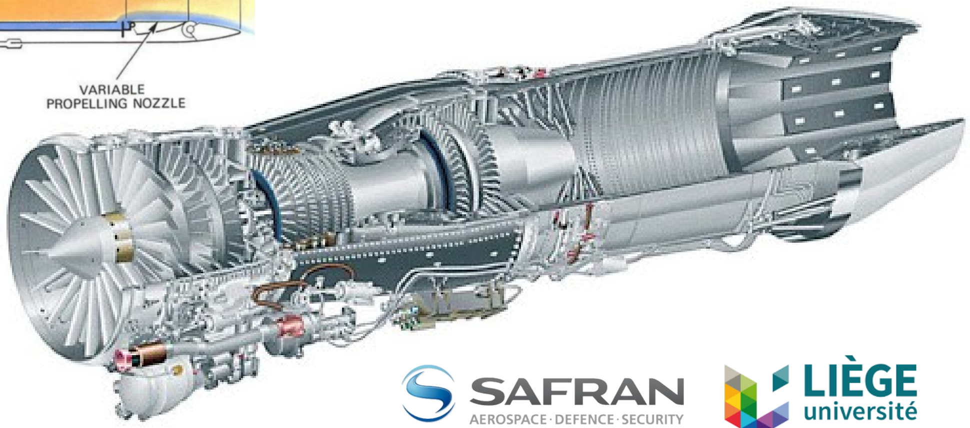
Garret/Honeywell TFE731
T ~ 1.5 – 2.5kN
BPR ~ 2.8

# 2. Jet engines

## *Afterburning turbofan: subsonic to supersonic flight*

- **Ram effect : 1’ - 1**
- **Core / primary flow**
  - LP + HP Compressor : 1 – 2
  - Combustion : 2 – 3
  - Expansion in turbine : 3 – 4
- **Bypass / secondary flow (below 1) :**
  - LP compression
  - Cooling of core
- **Mixer/Afterburner**
  - Mantle cooled by perspiration secondary flow
  - Mixing of primary and secondary flow
  - (Afterburning)
- **Exhaust by adjustable nozzle : 4’ – 5**



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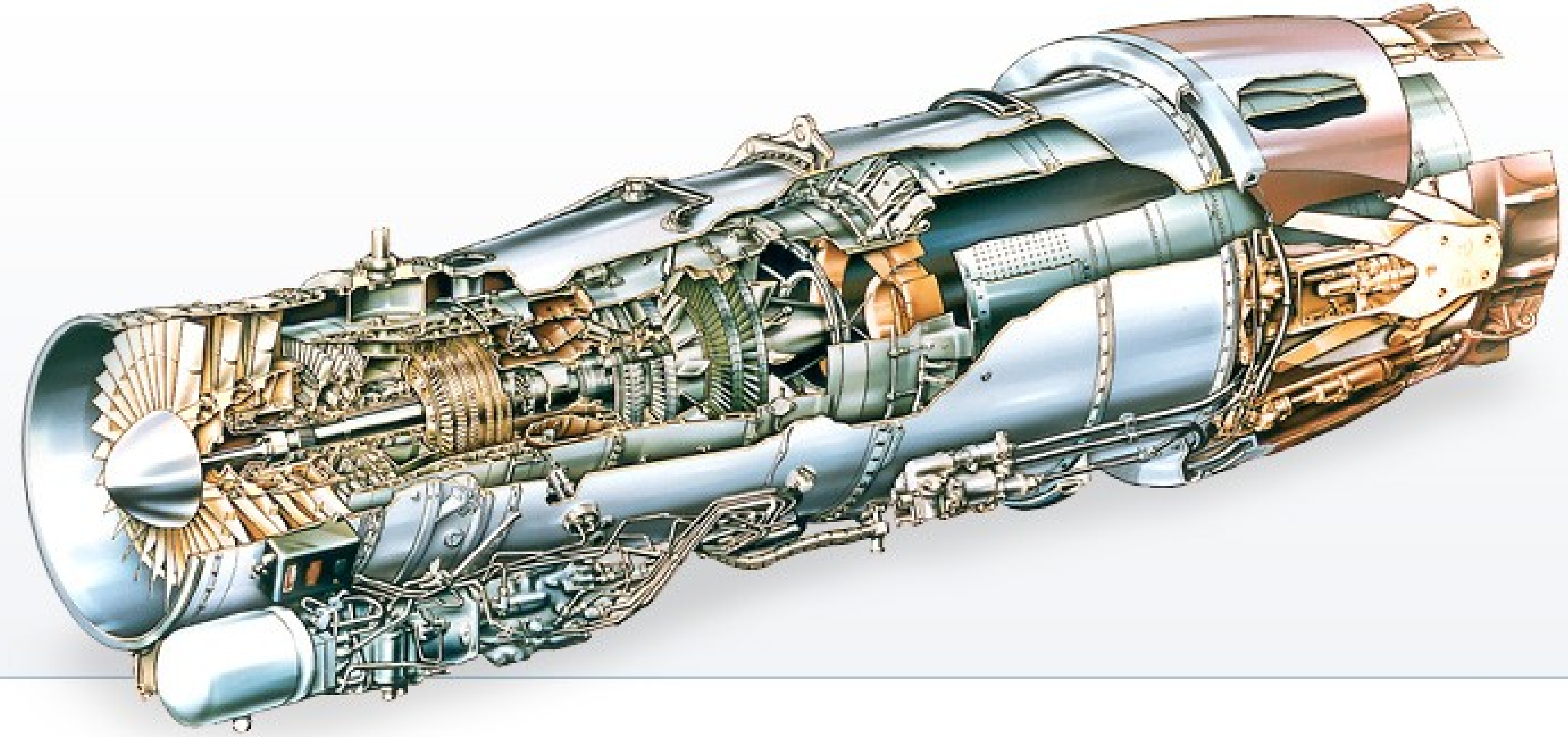
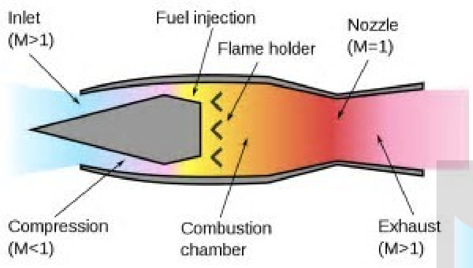
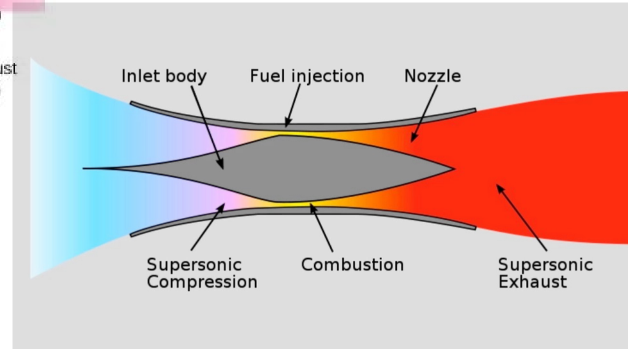
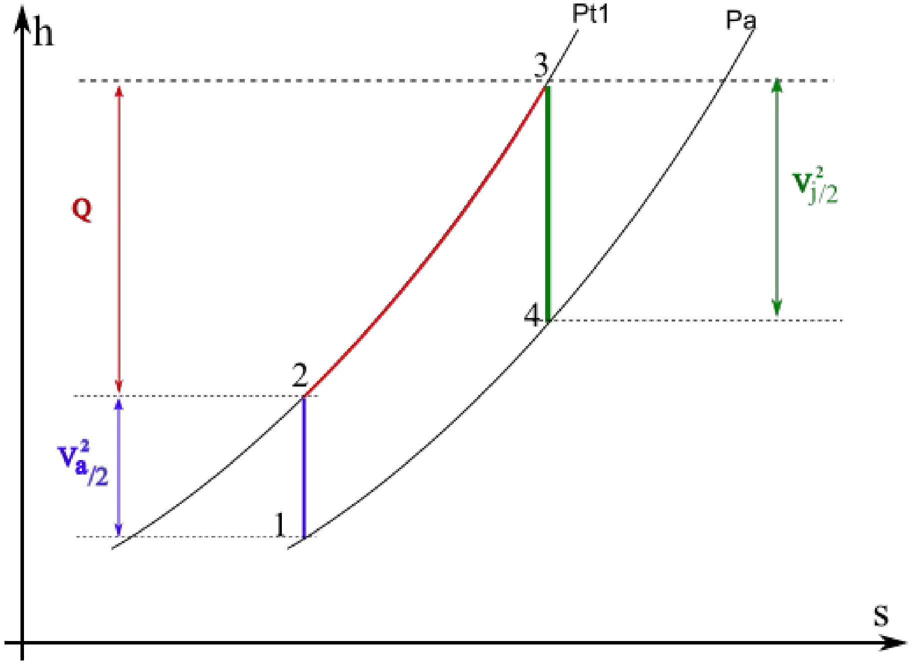
# Military turbofan w/ afterburning (J58)



- **Cycle without machines**
  - Convert kinetic energy va in pressure Pt by “RAM” effect : 1 - 2
  - Combustion : 2-3
    - RAMJET : subsonic combustion
    - Supersonic Combustion RAMJET
  - Expand in nozzle to form jet : 3 – 4

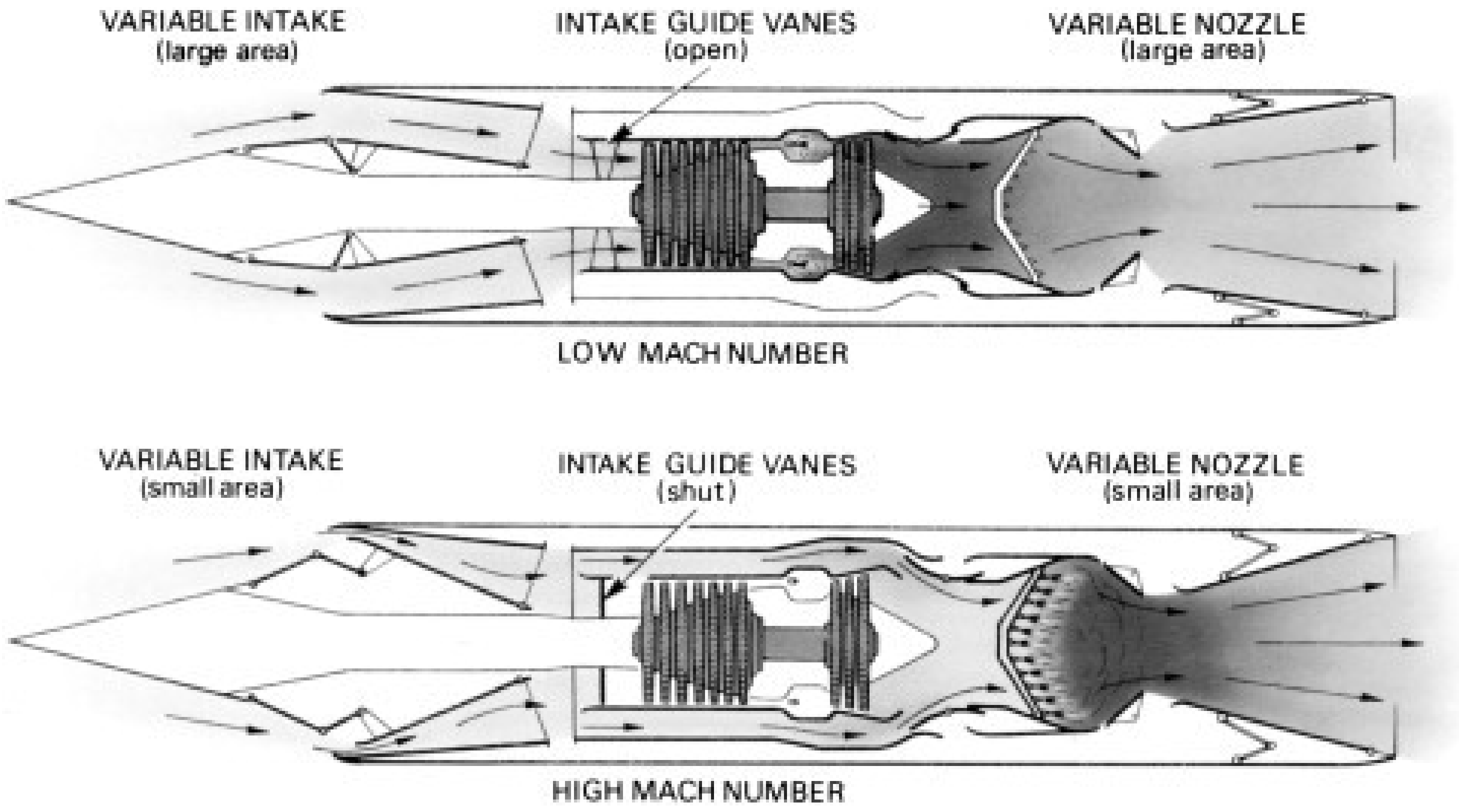


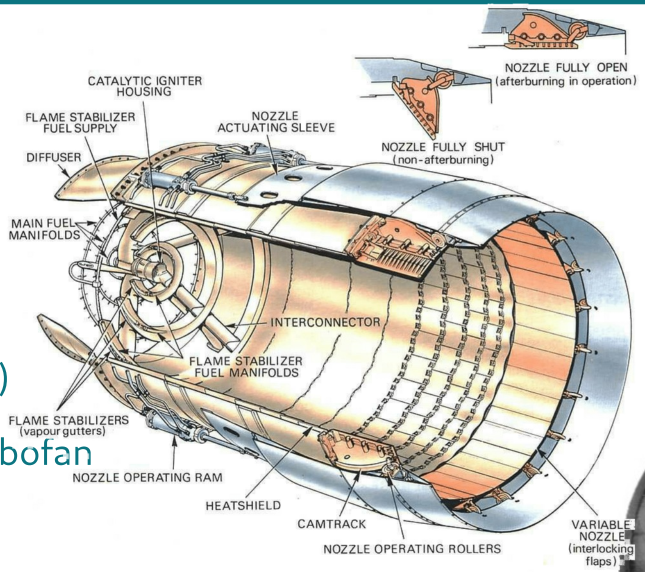



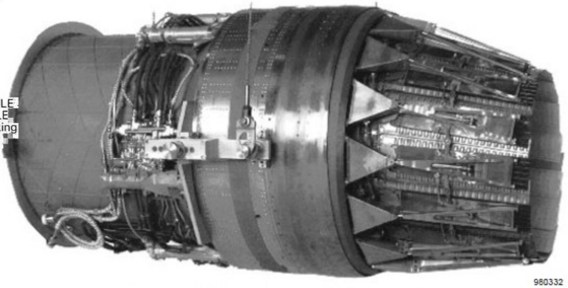
# 2. Jet engines

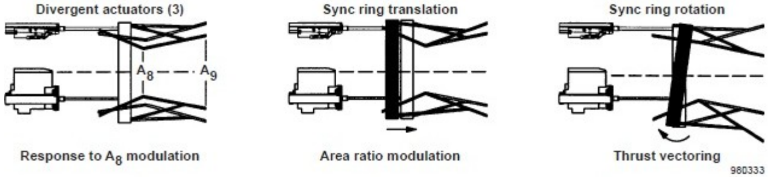
*Combined afterburning turbojet & ramjet*



- **Converging**
  - Sonic jet (wrt exhaust T) → lower speeds
  - Choked if NPR > NPR*
  - Adapted if NPR < NPR*
  - Fixed for civil turbofan (this could change …)
  - Variable throat for military/afterburning turbofan
- **Converging-diverging**
  - Higher speed range
  - Variable throat and expansion area to adapt to GG and flight
  - (thrust vectoring)
  - Afterburning turbojet/turbofan

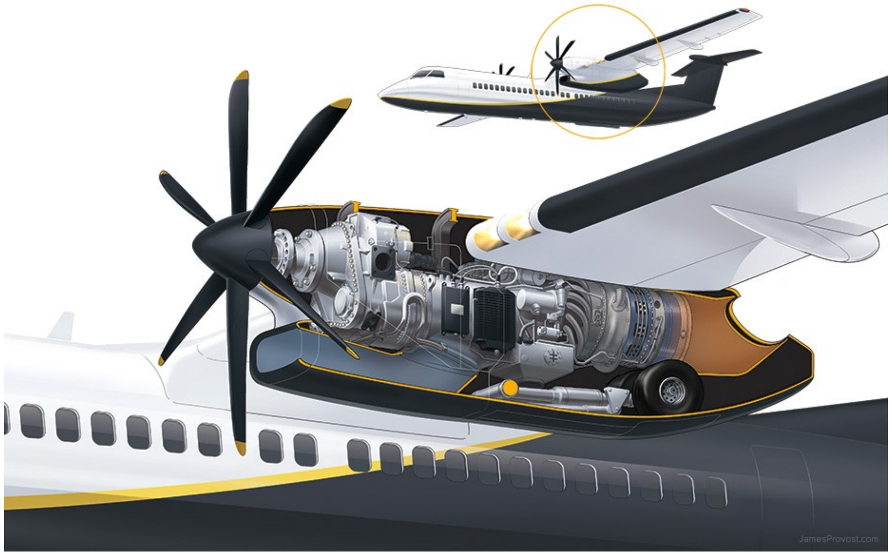


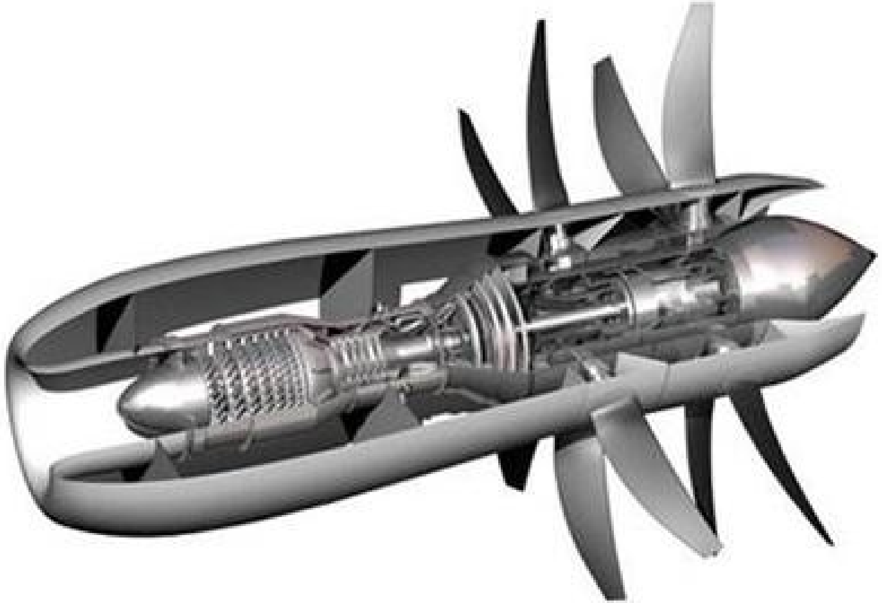



# 3. Propellers

*Mechanical acceleration by lift forces on propeller blades*





- **Accelerating / contracting stream tube**

$$\dot{m} = \rho v_0 S_0 = \rho v_1 S_1 = \rho v_2 S_2$$

- **Thrust** $T = \dot{m}(v_2 - v_0)$

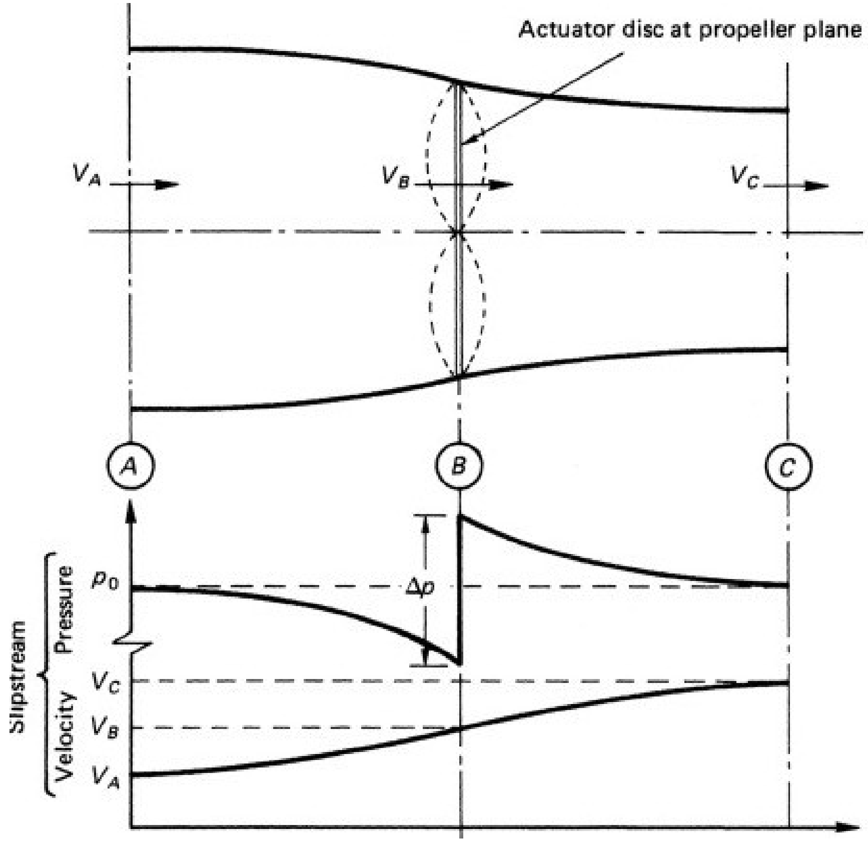
- **Propeller = actuator disk – pressure jump**

$$p_{1+} = p_{1-} + \Delta p$$

- **No losses up/down - Bernoulli**

$$p_{1+} = p_\infty + \frac{1}{2}\rho(v_2^2 - v_1^2)$$

$$p_{1-} = p_\infty + \frac{1}{2}\rho(v_0^2 - v_1^2)$$



- **Induction factor a**

$$v_{B+} = v_{B-} = v_B = v_A(1+a) = v(1+a)$$

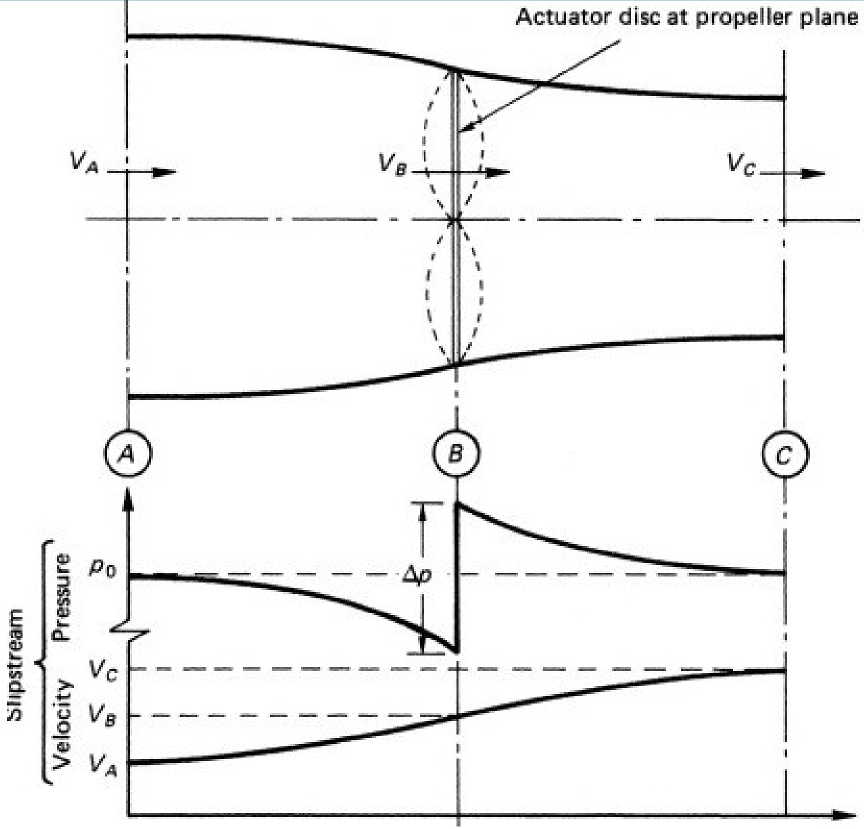
- **Thrust computed two ways**
  - stream tube control volume

$$T = \dot{m}(v_C - v_A) = \rho v_B S_1 (v_C - v_A)$$

  - pressure difference

$$T = (p_{B+} - p_{B-})S = \rho \frac{(v_C^2 - v_A^2)}{2} S = \rho \frac{(v_A + v_C)(v_C - v_A)}{2} S$$

$$\Rightarrow v_B = \frac{v_A + v_C}{2} \Rightarrow v_C = (1+2a)v_A$$



- **Thrust**

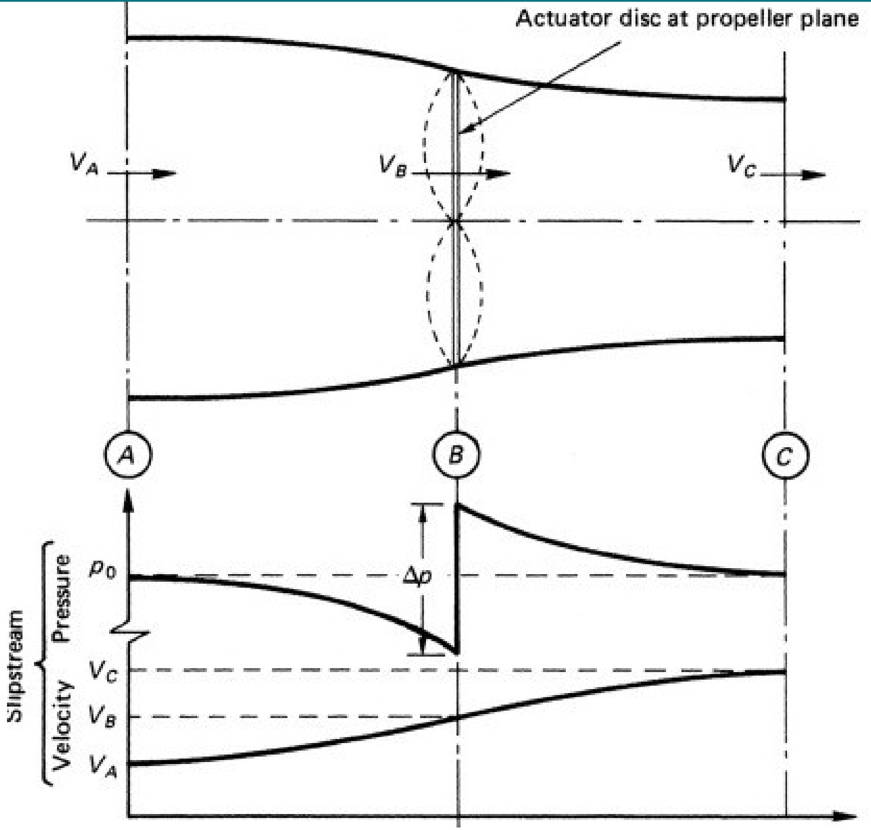
$$T = \dot{m}(v_C - v_A) = \rho v^2 S \cdot 2a(1+a)$$

- **Power**

$$P = \dot{m}\frac{1}{2}\left(v_C^2 - v_A^2\right) = \rho v^3 S \cdot 2a(1+a)^2$$

- **Propulsive efficiency**

$$\eta_p = \frac{Tv}{P} = \frac{1}{1+a}$$

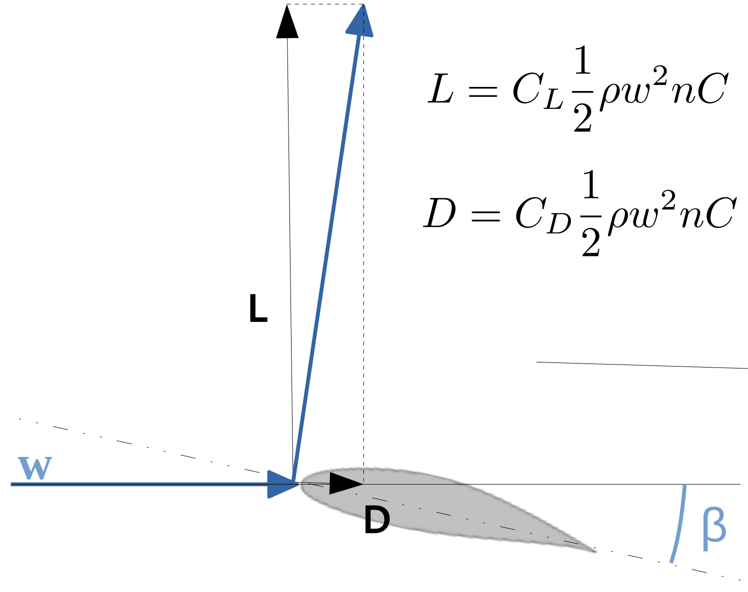


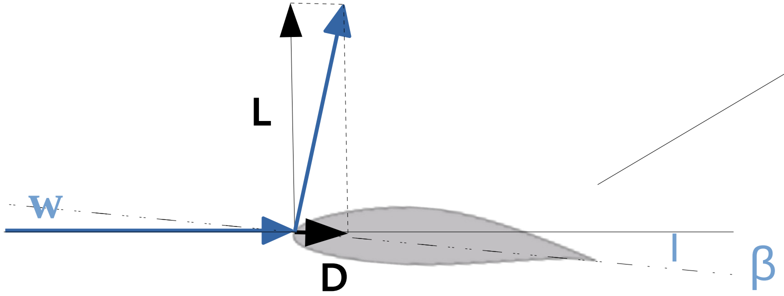
# 3. Propellers

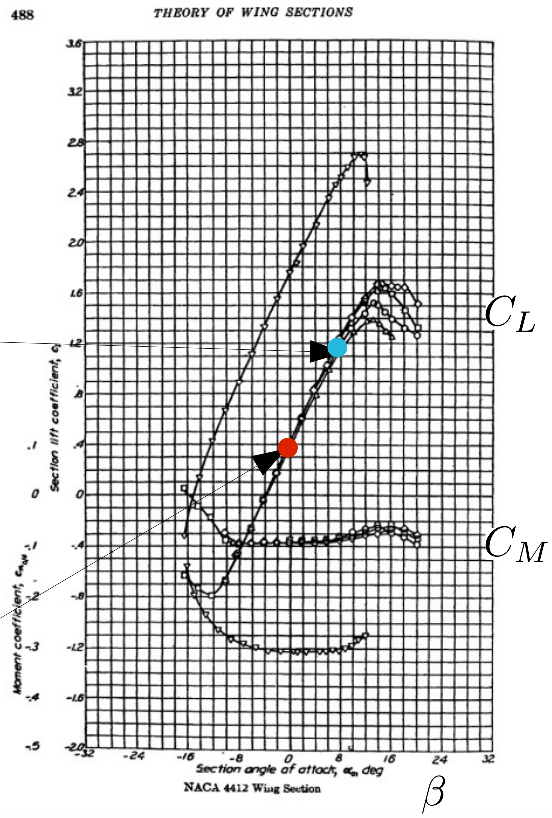
*Blades : airfoil lift/drag polars*

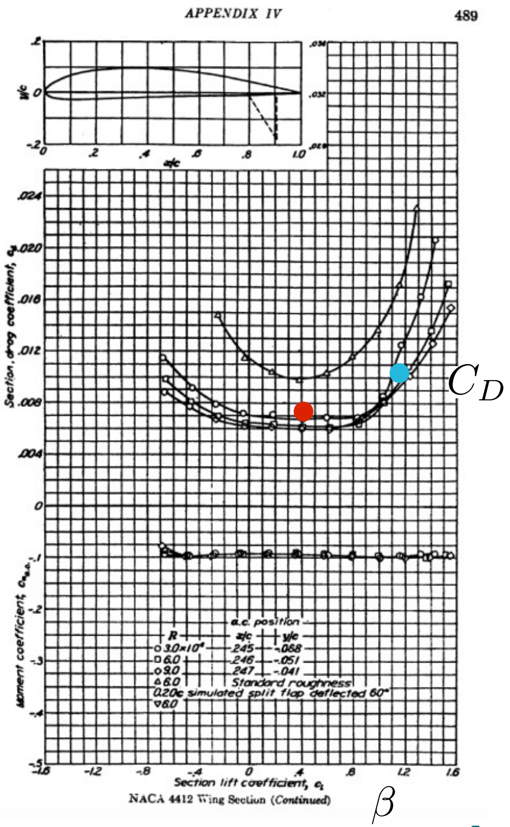
$$L = C_L \frac{1}{2} \rho w^2 n C$$

$$D = C_D \frac{1}{2} \rho w^2 n C$$



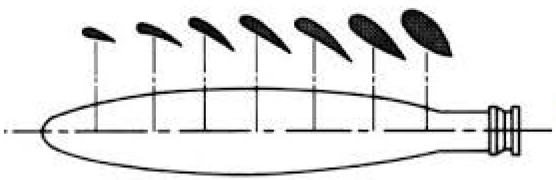



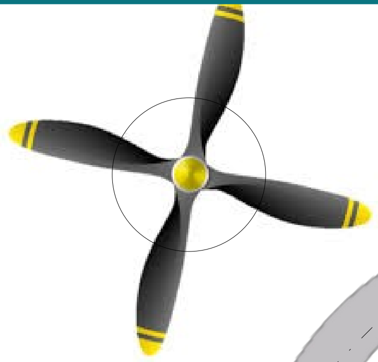



# 3. Propellers

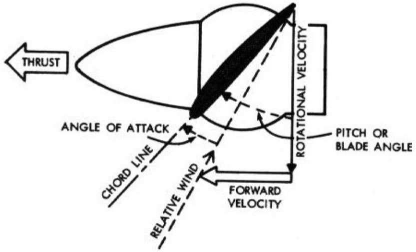
## *Blades – layout and operating parameters*

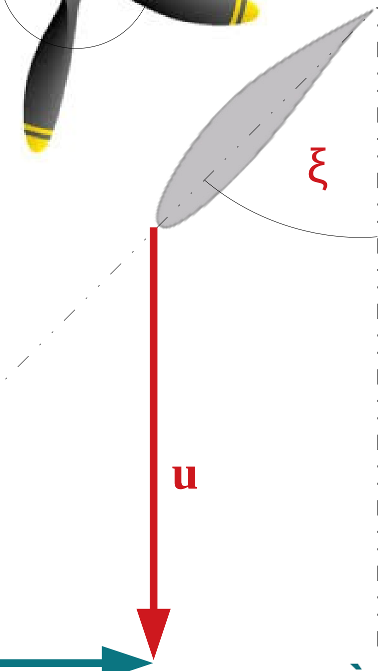
- **Geometry**
  - Tip radius $R_t$
  - (Radial distribution of) blade profile
  - Radial distribution of chord $C(r)$ / Solidity $\sigma(r) = N\,C(r)/2$

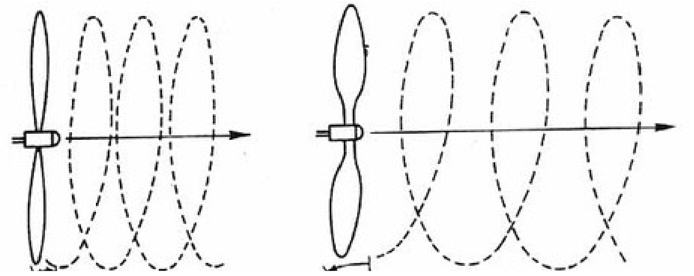




- **Operating parameters**
  - *Stagger* or *pitch* angle $\xi$
  - Rotation speed $\Omega$ → local blade speed $u = \Omega r$
  - *Advance ratio* $J = v_a/u_t = v_a/\Omega r_t$



# 3. Propellers

## *Blades : forces on blade element section*

- **Induction factor a → v(1+a)**
- **Relative velocity w, flow angle β**

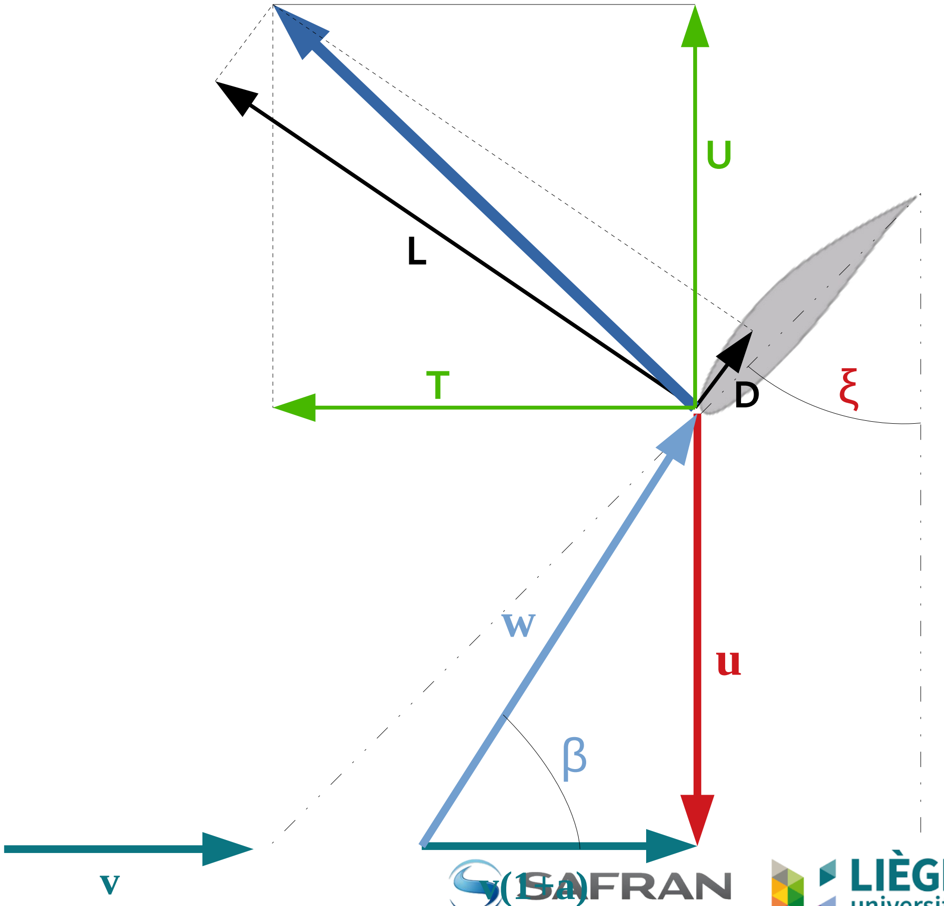
$$\mathbf{w} = (1+a)v_0\mathbf{e}_x + u\mathbf{e}_y$$

- **Lift/drag forces L & D as a function of**
  - Relative velocity w
  - Incidence i = ξ - β
- **Compute thrust T and tangential force U**

$$T = L\sin\beta - D\cos\beta$$

$$U = L\cos\beta + D\sin\beta$$

- **Recompute induction factor a from T**



- **Thrust coefficient**

$$T = \dot{m}(v_C - v_A) = \rho v^2 S \cdot 2a(1+a) \qquad \Rightarrow C_T = \frac{T}{\rho S u_b^2} = J^2 \cdot 2a(1+a)$$

- **Power coefficient**

$$P = \dot{m}\frac{1}{2}\left(v_C^2 - v_A^2\right) = \rho v^3 S \cdot 2a(1+a)^2 \qquad \Rightarrow C_P = \frac{T}{\rho S u_b^3} = 2a(1+a)^2$$

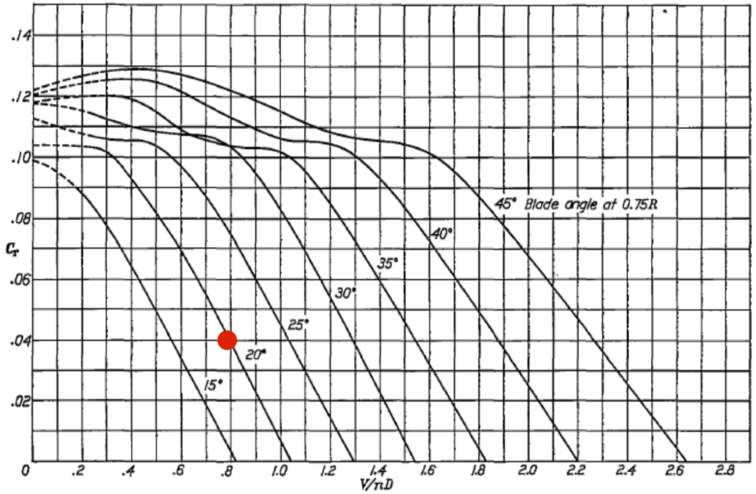



FIGURE 3.—Thrust-coefficient curves for propeller 5868-9, Clark Y section, 2 blades.

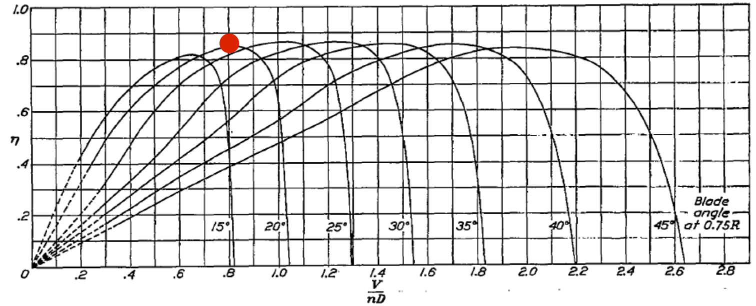
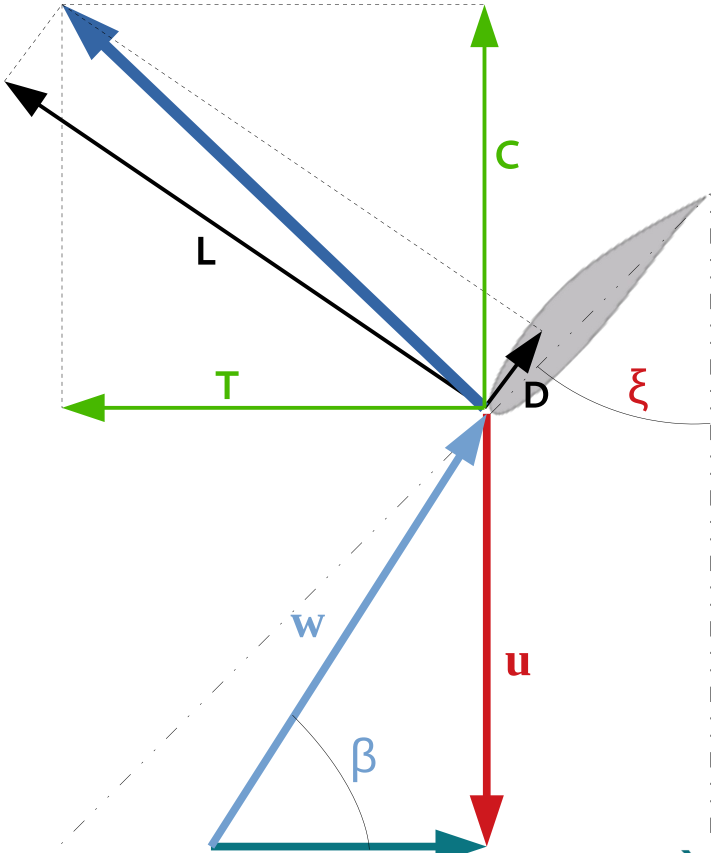



FIGURE 5.—Efficiency curves for propeller 5868-9, Clark Y section, 2 blades.



*Operation point: variation of advance ratio (flight speed or rotation speed)*

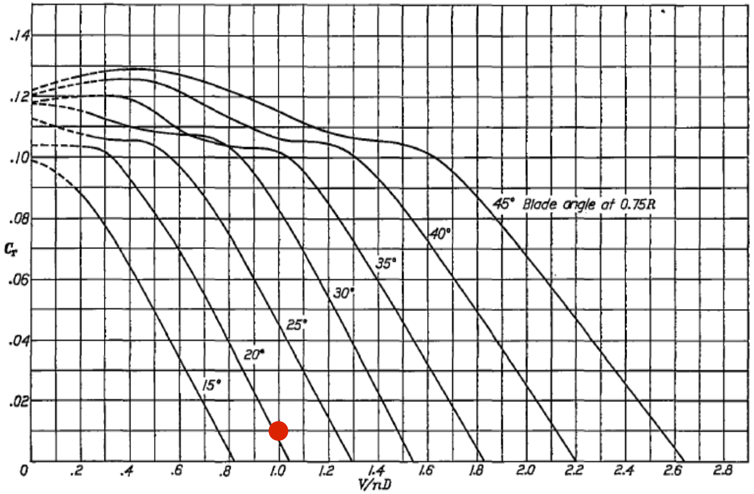



FIGURE 3.—Thrust-coefficient curves for propeller 5868-9, Clark Y section, 2 blades.

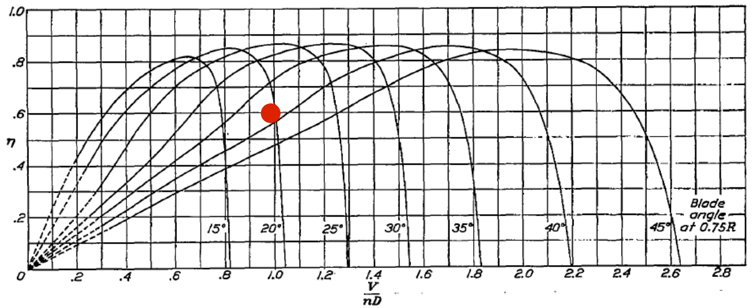
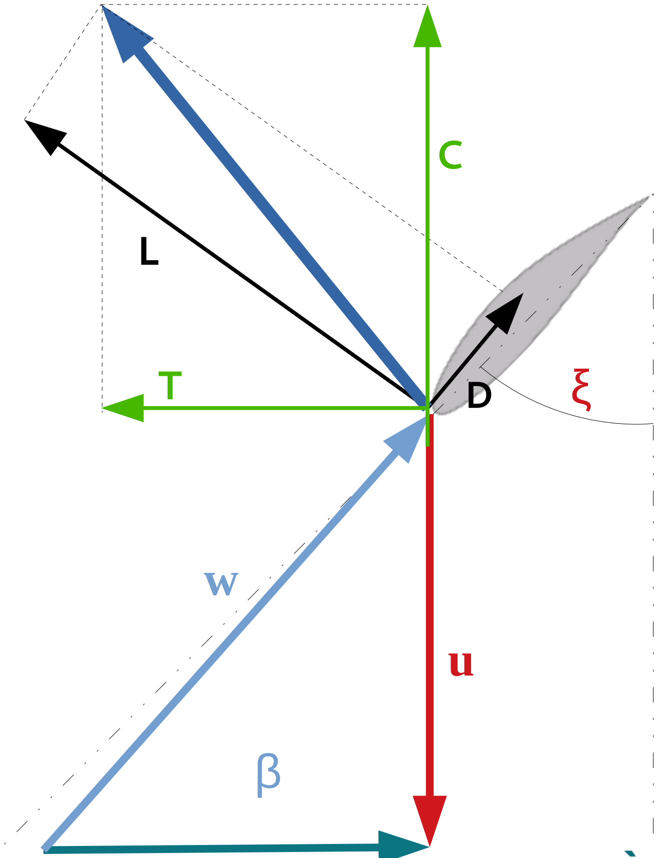



FIGURE 5.—Efficiency curves for propeller 5868-9, Clark Y section, 2 blades.



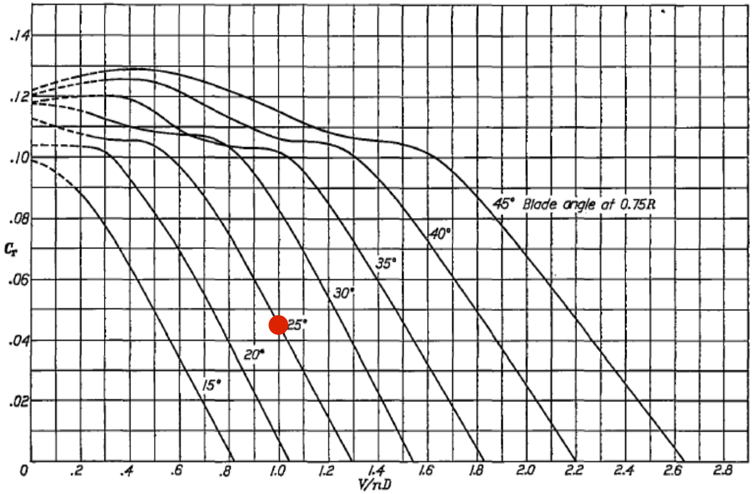



FIGURE 3.—Thrust-coefficient curves for propeller 5868-9, Clark Y section, 2 blades.

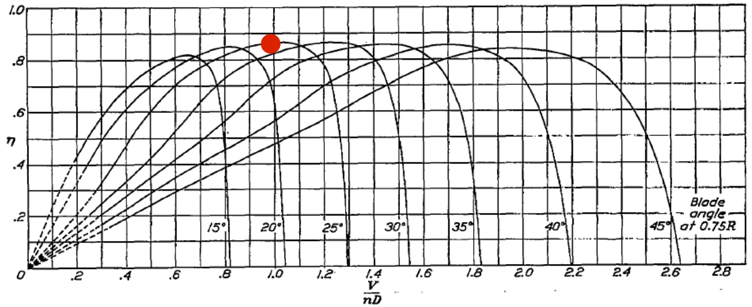
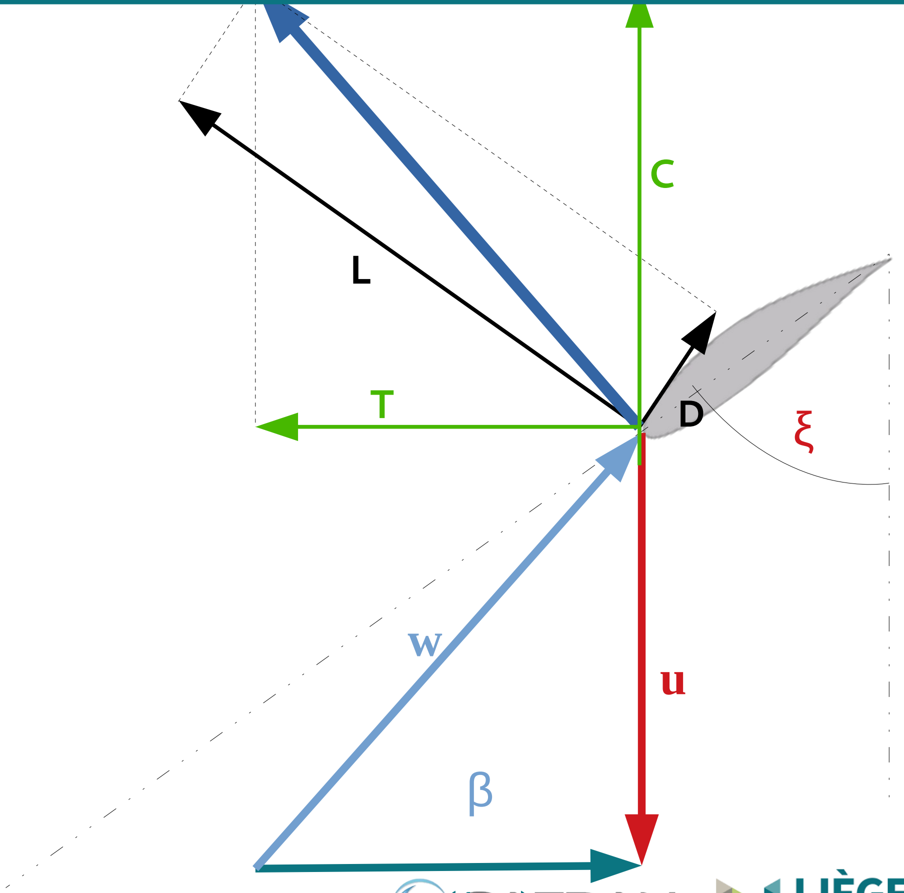



FIGURE 5.—Efficiency curves for propeller 5868-9, Clark Y section, 2 blades.



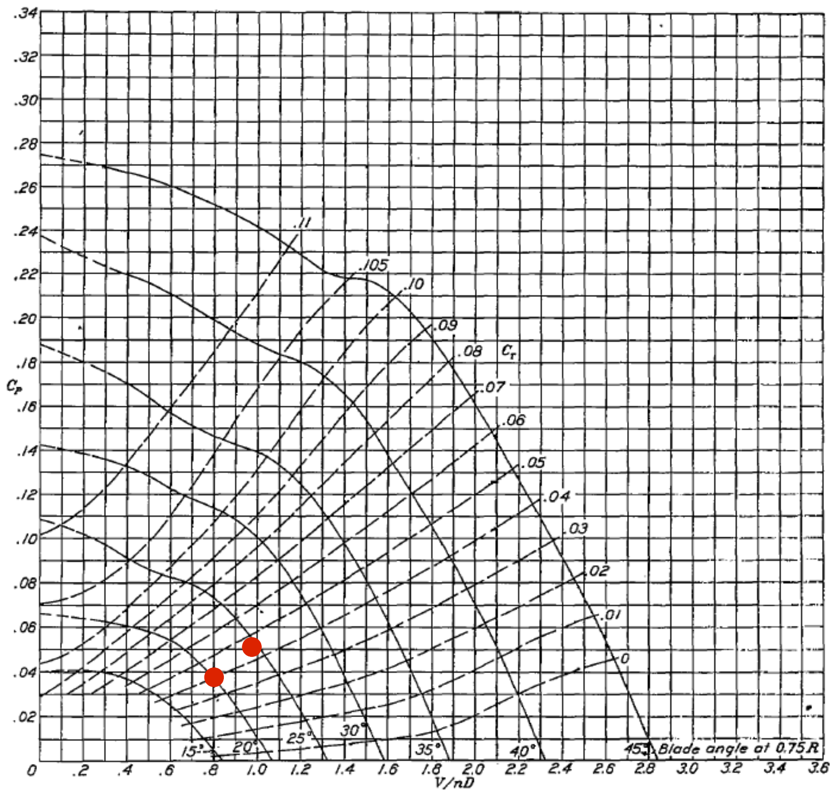
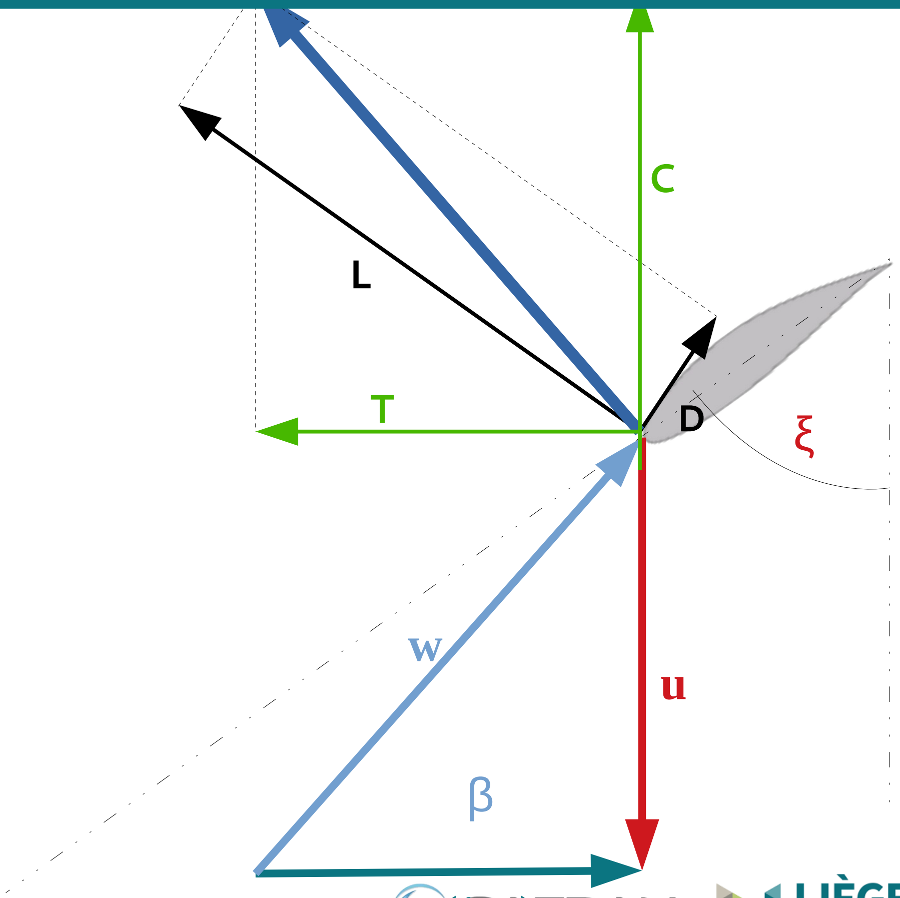
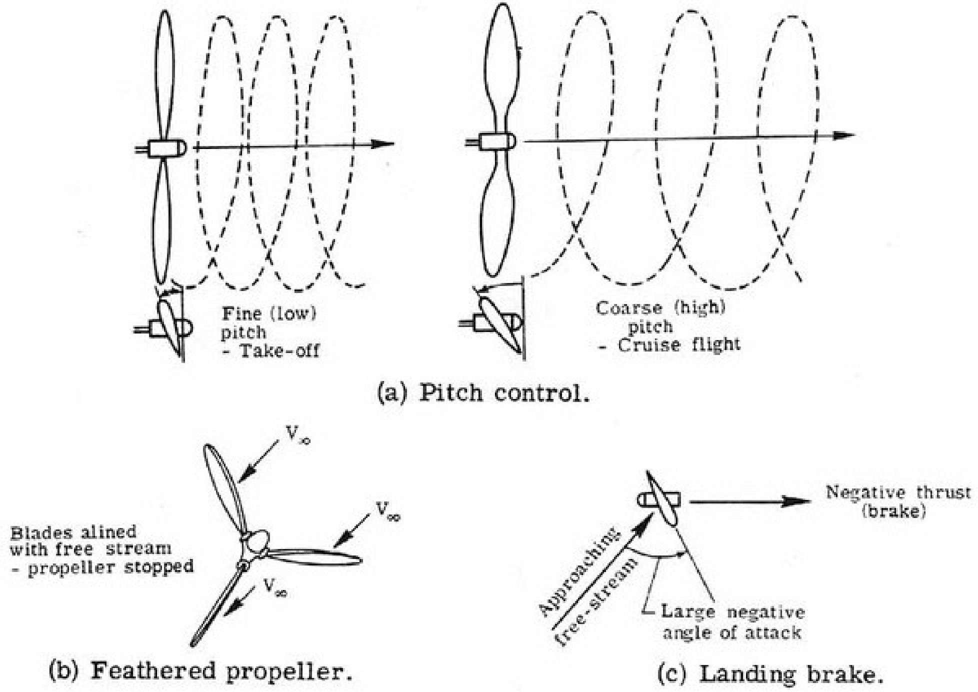



FIGURE 6.—Power-coefficient curves for propeller 5868-9, Clark Y section, 2 blades.



# 3. Propellers
*Control*

- **Means of control**
  - Engine power / rotation speed : $T \sim n^2$, $P \sim n^3$
  - Advance speed vs rotation speed : "gear box"
    - Fine pitch ~ low gear :
      - high thrust at low advance speed : take-off, taxi, ...
      - limited flight speed
    - Coarse pitch ~ high gear :
      - low thrust at take-off
      - higher air speeds

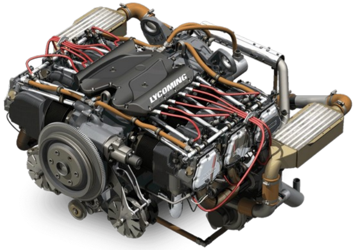



- **Propeller types**
  - Fixed pitch / ground adjustable ~ single gear
  - Variable pitch propellers
    - Inflight adjustable: change throttle and pitch angle independently ~ manual gear box
    - Fixed velocity: governor adjusts pitch to keep constant rotation speed ~ automatic gearbox

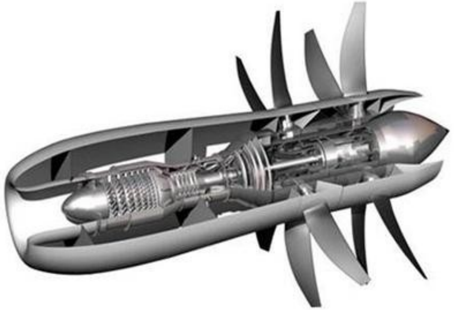
# 3. Propellers

*Power generators*

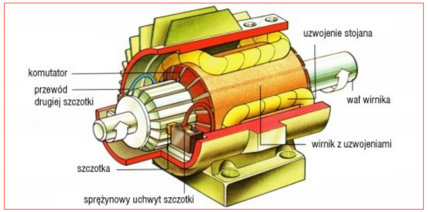
- Internal combustion engine




- Gas turbine: turboprop & CROR

- Future of propulsion systems : electric or hybrid, distributed …

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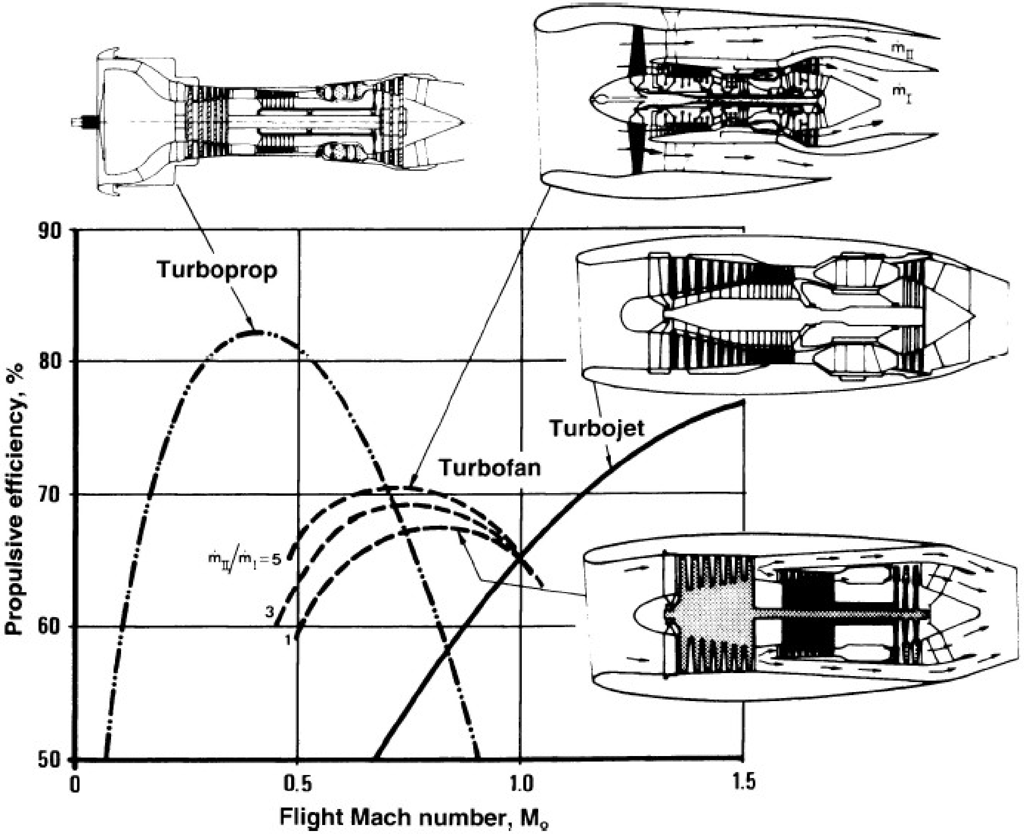
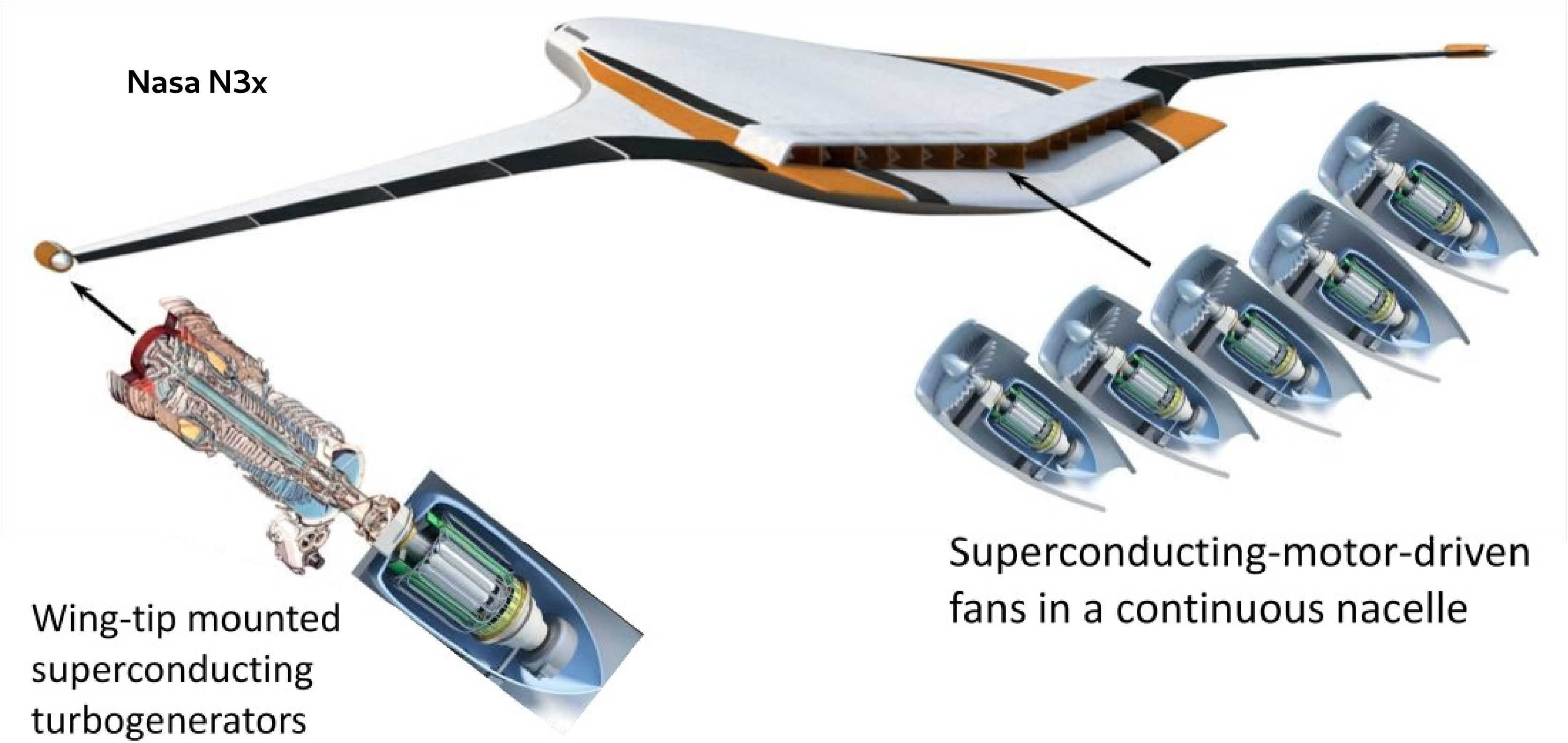
# 4. Choosing the propulsion system

- **Typical : cruise speed**
  - Propeller : 0.1 < Ma < 0.7
  - High BPR turbofan : 0.7 < Ma < 1
  - Low BPR turbofan / turbojet : Ma > 1
  - RAMJET : Ma > 2
  - SCRAMJET : Ma > 5
- **Extension of operating range of propellers to transonic**
  - (Variable pitch turbofans)
  - Transonic open rotors (CROR)
  - Ducted fan for hybrid propulsion



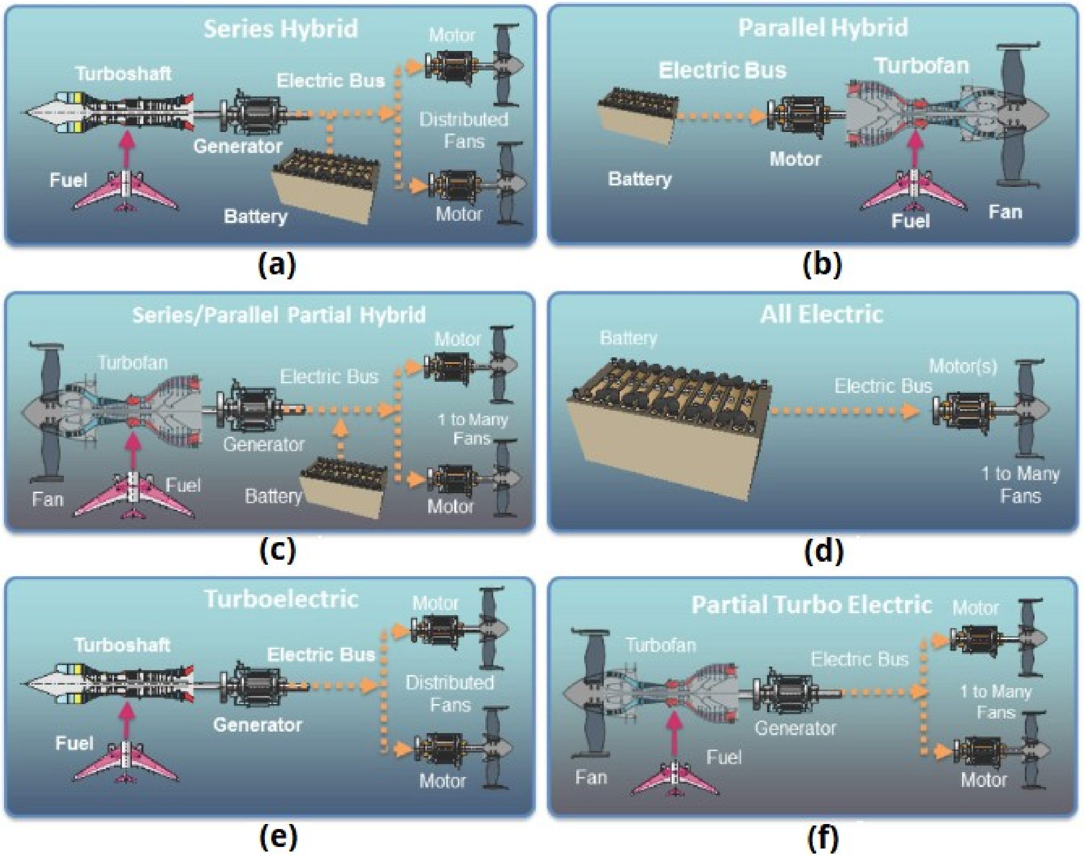
# 4. Choosing the propulsion system

*Future: integrated / hybrid / distributed propulsion systems ?*



# 4. Choosing the propulsion system

*hybrid / new propulsion stems ?*





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